Technical Report 1018

Infrared Detections of Satellites with IRAS

E.M. Gaposchkin R.J. Bergemann

26 September 1995

Lincoln Laboratory

MASSACHUSETTS INSTITUTE OF TECHNOLOGY

LEXINGTON, MASSACHUSETTS



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INFRARED DETECTIONS OF SATELLITES WITH IRAS

AND STREET

E.M. GAPOSCHKIN
Group 91

R.J. BERGEMANN
Consultant

TECHNICAL REPORT 1018

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LEXINGTON

ABSTRACT

The sky survey made by the infrared astronomical satellite (IRAS) in 1983 included observations of artificial earth satellites. The data base (with celestial objects removed) was correlated with the NORAD space catalogue to identify 452 satellites in orbit above the IRAS 900-km altitude. The flux density in three of the four wavelength bands has been analyzed to determine the temperature, emissivity, and absorptivity of the identified resident space objects.

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1. INTRODUCTION

Optical observations of artificial earth satellites have been made since the launch of Sputnik. Early measurements made using reflected visible light [1] were for information about the satellite position. Later, photometric observations were made to investigate properties of the satellite. Initially, signature data were obtained in the form of a time series or light curve. Later, calibrated photometric observations were used to discuss the reflective properties, such as reflectivity, and configuration of satellites [2]. Observations of the self-emitted IR radiation from satellites were also obtained. IR data are complementary to the visible band and UV data and can provide information about the absorptivity and emissivity of satellite materials. These ground-based observations were frustrated by large and variable atmospheric absorption [3] in addition to sensor calibration problems. Recently, high-quality ground-based IR satellite observations have been made [3].

The infrared astronomical satellite (IRAS) was launched in 1983 to perform an all-sky survey of the IR portion of the spectrum. While stellar objects—stars, nebulae, comets, and asteroids—were the primary objects of interest, IRAS also observed satellites and space debris. A number of attempts have been made to extract these data and demonstrate that IRAS data can be used to characterize IR emission of orbital debris. De Jong and Wesselius [4] and Anz-Meador et al. [5] used unprocessed IRAS detector data to search for debris. These studies found a few moving sources from searching only a few days of the mission data. Dow [6,29] used objects found in the Sky Brightness Images over the entire IRAS mission. All studies concluded that IRAS had the capability to observe quite small debris—Anz-Meador claimed sizes down to 1 mm—but none systematically attempted to isolate data on known satellites.

The Astronomical Group at Groningen reprocessed all the raw IRAS data tapes with the objective of finding all satellite and debris detections [4,7,8]. This effort produced a debris data base containing more than 190,000 detections. As a first step in analyzing this data base, we have concentrated on the catalogue of known satellites in 1983. By correlating positions in the debris data base with the known satellite catalogue, we have extracted the true satellite detections and been able to discuss the associated radiometric data. This analysis is the subject of this report.

The IRAS satellite made many observations of artificial earth satellites. Processing for astronomical objects (stars, planets, comets, and asteroids), Beichman [9] eliminated all satellite and debris data. The Astronomical Group at Groningen reprocessed all the raw IRAS data tapes. They abandoned the original approach of hours, days, and weeks confirmation, as this method was intended to eliminate observations of objects that did not reappear at the same place in the sky—just those objects of interest here. They established a method that examined each detection for multiple detector "hits" that were consistent in timing of their passage across the focal plane. The approach was very permissive, having wide bounds for accepting a plausible detection. The final catalogue contained more than 190,000 detections. The flux values were limited to those greater than 0.1 Jy (W/m²/Hz). Also, they did not consider data from the $100-\mu$ band. This data base was provided to Lincoln Laboratory. Consideration was limited to those detections with a signal-to-noise ratio (SNR) greater than about three: a data base of about 136,000 detections.

2. IRAS INSTRUMENT

The IRAS operated from March 1983 until cryogen depletion in November 1983. IRAS was in a sun-synchronous orbit with the orbital plane (800-km altitude, 99.2° inclination) nearly perpendicular to the sun line, as illustrated in Figure 1. The telescope was always pointed away from the earth center, i.e., toward the zenith. The spacecraft attitude control system allowed pole-to-pole scans in ecliptic longitude from 60° to 120°; most scans were taken between 84° and 96°.

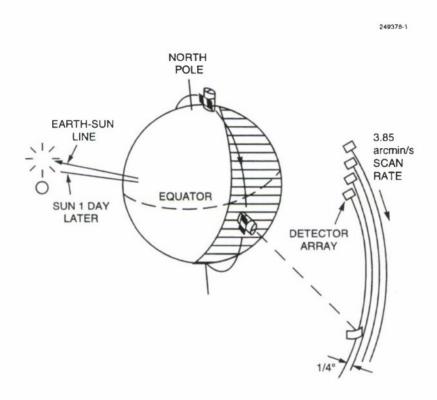


Figure 1. IRAS orbit geometry.

The IRAS detectors covered four long wavelength IR bands having flux density reporting values of 12, 25, 60, and 100 μ . These correspond to peak flux densities of 242, 116, 48, and 29 K. The peak wavelength for the nominal free-space temperature of 300 K, $T = 2898/\lambda$, is 9.66 μ near the lower cutoff of the 12- μ band.

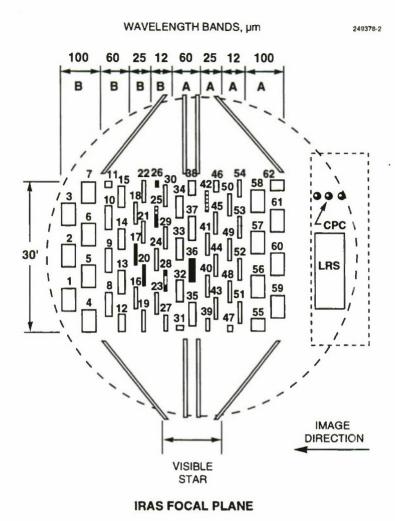


Figure 2. IRAS focal plane.

The IRAS detectors in each of the bands were arranged on a 30-arcmin focal plane, as illustrated in Figure 2. A star or satellite crossing the scanning focal plane would traverse a number of detectors in several wavebands. Each of the 59 detectors had a different response function, as tabulated in Beichmann, Table II.C.5, page II-18. The detectors had a rectangular shape: 0.75×4.5 arcmin for the 12- and $25-\mu$ detectors and 3×5 arcmin for the 60- and $100-\mu$ detectors. As shown in Figure 2, the spacecraft attitude was such that in the scan direction, motion of fixed stellar sources or satellites across the focal plane was along the short detector axis. Consequently, the positional accuracy in the along-scan direction was much better.

The camera electronics used a 6-Hz lowpass filter to "despike" the data. The detectors were sampled at 16 Hz. The scanning speed, determined by the IRAS orbital motion, was 3.85 arcmin/sec. Therefore, a stationary source traversed the 0.75-arcmin detector in 0.19 sec, giving about three samples for a detection. Consequently, all objects (stellar sources, solar system objects, satellites, and debris) have very similar detections. This is illustrated in Figure 3 (from Wesselius et al.). Therefore, the response function is not a suitable vehicle for identifying satellites and debris.

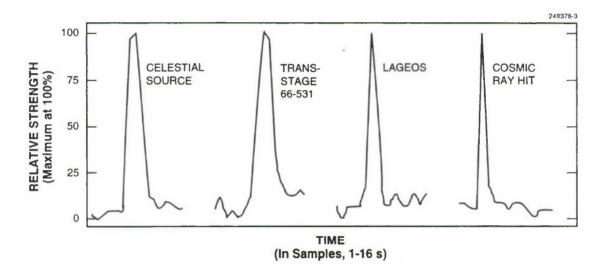


Figure 3. Response due to celestial objects.

The baseline IRAS method of seconds, hours, days, and weeks confirmation for object identification of a detection relies on the object being stationary. Since rapidly moving satellites and debris would be screened out, this method was abandoned. Wesselius et al. used the following approach. Satellites at a geometric range greater than 10,000 km will have an absolute velocity between 0.25 arcmin/sec (geosynchronous) and 10 arcmin/sec and apparent velocity between 4 and 10 arcmin/sec. The object can take more than 10 sec to cross the IRAS focal plane and will cross each detector at a different time, as illustrated in Figure 4. Here, the detector samples are displayed, as a function of time, for detectors arranged as they were on the focal plane. By correlating the time history of detections, the RSO and debris hits were selected from the raw data. From this data one obtains the amplitude of the hit in each detector, the position, and relative velocity of the object. Because of the relatively inaccurate position, the relative velocity is particularly important in correlating the hits with a catalogue of known satellite orbits.

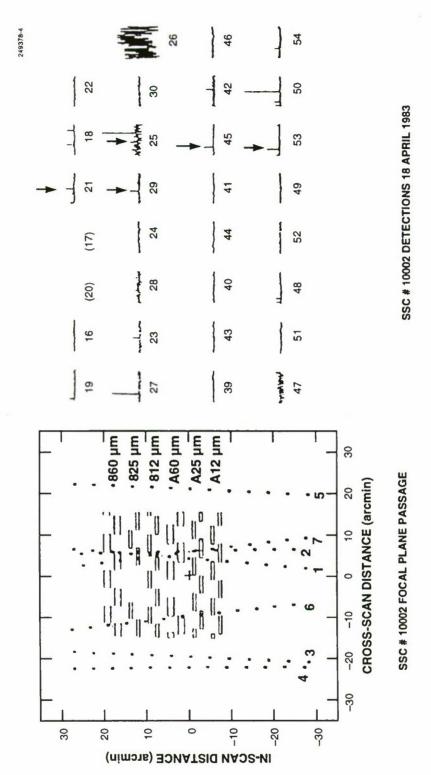


Figure 4. SSC 10002 focal plane passage and detections.

3. ANALYSIS CONSIDERATIONS

The IRAS radiometric data is provided in Janskys uncorrected for color. The flux in-band was converted to Janskys "without prejudice" by assuming a λ^{-1} frequency distribution of the objects' radiation. This allowed processing the data without making assumptions about the temperature or wavelength distribution of the source, and the bands could be processing independently. Data reduced in this way is said to be uncorrected for color. The *IRAS Explanatory Supplement*, page II-27, describes how to make the color correction and obtain "true" observed Janskys. That procedure is implemented here using the tables given on page VI-26. Note that these tables can, in principle, be derived from the spectral response function, Table II.C.5, page II-18. These tables have been reconstructed to two significant figures. The source of disagreement is unknown, but it probably derives from subtleties in the implementation of the calibration process. The basic process for making the color correction is as follows.

The observed flux density, at a reference wavelength λ_0 , for a source at temperature T, is

$$F_{obs}^{\lambda_o}(T) = \frac{\epsilon A}{\pi r^2} \mathcal{F}_{\lambda_o}(T) \qquad \left(\frac{W}{m^2 \mu}\right) \quad ,$$

where the Planck flux density is

$$\mathcal{F}_{\lambda}(T) = \frac{3.74185 \times 10^{8}}{\lambda^{5} \left(e^{\frac{14388.3}{\lambda^{T}}} - 1\right)} \qquad \left(\frac{W}{m^{2}\mu}\right) ,$$

and λ is in microns (Allen), A is the projected area of the object, ϵ is the emissivity, and r is the range to the object. The IRAS data is reported in Janskys, viz., 10^{-26} W/m²/Hz. Using the relation that $f\lambda = 2.99792458 \times 10^{14}$ µ/sec, one can convert F_{obs}^{λ} to Janskys with

$$J_{obs}^{\lambda_o}(T) = F_{obs}^{\lambda_o}(T) \left(\frac{\lambda_o^2}{2.99792458 \times 10^{-12}} \right) \left(\frac{W}{m^2 Hz} \right)$$

The temperature of an object can be found as follows. The temperature dependence of the observed flux $J_{\text{obs}}^{\lambda}(T)$ depends on λ_0 . The observed flux also depends on ϵ , λ , and r. Assuming ϵ is independent of λ , the ratio of observations made at the same time

$$\Re(T) = \frac{J_{obs}^{\lambda_1}(T)}{J_{obs}^{\lambda_2}(T)}$$

for bands λ_1 and λ_2 is a monotonic function of T. Given a pair of true flux densities, the temperature is obtained from this function. Now we define the color correction, K_{λ} , to convert the quoted, i.e., reported, flux density $f_{\lambda} = J_q^{\lambda}$ to the "true" observed flux density as

$$J_{obs}^{\lambda_o} = J_q^{\lambda_o} / K_{\lambda_o}$$

or converting from Janskys to $W/(m^2\mu)$ then

$$F_{obs}^{\lambda_o} = F_q^{\lambda_o} / K_{\lambda_o}$$

The ratio

$$\Re_{q}(T) = \frac{J_{q}^{\lambda_{1}}}{J_{q}^{\lambda_{2}}}$$

and the $K_{\lambda}s$ are also monotonic functions of T. They can be calculated from the spectral response function for each band. This calculation is described in the IRAS Explanatory Supplement, page VI-27. The temperature from the flux density ratios f_{12}/f_{25} and f_{25}/f_{60} can be obtained. The temperature obtained from the quoted flux density ratios is the same as that obtained from the color-corrected "true" observed flux densities.

Initial modeling of RSO flux and temperature for experiment planning is done assuming that the solar absorptivity α is equal to the LWIR emissivity ϵ . While this is appropriate for solar-cell-powered payloads of principal interest, many objects, particularly debris, may have surfaces with the value of absorptivity different from the emissivity, resulting in a large range of free-space temperatures.

To aid in the discussion a simple model for the radiant flux is developed. This calculation requires a number of assumptions. First, the object is in thermal equilibrium, and all the absorbed solar radiation energy is reemitted from each element following Lambert's law. Consequently, we ignore the energy that is reradiated as microwave radiation. Second, we assume there is no contribution for earth-upwelling radiation. Third, we assume that the satellite has one surface material, which absorbs solar radiation and radiates at a single temperature. Finally, we assume that the satellite is not in or near eclipse. We have developed four simple models, which are listed in Table 1. These constitute four idealized satellite configurations.

TABLE 1
Free-Space Temperature for Basic Satellite Shapes

Principal Shape	Temperature	n= A _s /A _p
Sphere	278 K	4
Cylinder (Radius_r, Length L)	295 K	π(1+ r/L)
Flat Solar Panel (Two Sides)	330 K	2
Flat Solar Panel (One Side)	393 K	1

The free-space equilibrium temperature is determined by setting the solar power absorbed by the projected area equal to the total power reradiated by the surface area according to the Stephan-Boltzmann expression, in MKS units;

1360
$$\alpha A_p = 5.6692 \times 10^{-8} \in A_s T^4$$
,

where A_n is the projected area and A_n is the surface area. This relation then gives

$$T = (393 \text{ K}) \left(\frac{A_p}{A_s}\right)^{\frac{1}{4}} \left(\frac{\alpha}{\epsilon}\right)^{\frac{1}{4}} .$$

For these simple models using the solar radiation input of 1360 W/m², one can calculate the free-space temperature $T_{\alpha} = \epsilon$, assuming that the absorptivity α equals the emissivity ϵ . If the absorptivity is greater/lesser than the emissivity, the temperature will be greater/lesser than this calculated temperature. From the observed temperature and emissivity, we can obtain the absorptivity from

$$\alpha = e \left(\frac{T}{T_{\alpha = e}} \right)^4$$

The expression for total radiant intensity is then

watts /steradian =
$$\frac{1360 \quad \alpha A_p}{n \quad \Pi}$$
,

where $n = A_s/A_p$, as given in Table 1. Thus, the total radiant intensity for a sphere is $108 \alpha A_p$ watts/steradian. This simplified model requires that the object's radiance vs. wavelength be a Planck function with LWIR emissivity ϵ independent of wavelength. For objects in near-earth orbit, the effects of earth temperature, earth-reflected sunlight, and earthshadowing must be included in the RSO temperature estimate [12].

¹ For this analysis, the value of the solar constant, 1360 W/m², is taken from Allan, page 169 [10]. Recent measurements from Nimbus-7 suggest a value of 1372 W/m² for 1983 [13].

4. SPACECRAFT MATERIALS

All model calculations depend on the physical properties of the satellite surface materials. For high-fidelity modeling, detailed information is necessary. Because this information is unavailable, illustrative values are used, as compiled by Dow [6] and reproduced in Table 2 for α and ϵ . Also given are temperatures for a sphere² (T1) and flat plate (T2) covered with the material. The values of α/ϵ range from 0.1 to 10.0. Figure 5 plots reflectivity ($\rho = 1 - \alpha$) vs. the tabulated α/ϵ ratios. As seen, almost all values of α and ϵ are possible. For example, the temperature of a gold-plated sphere can rise above the nominal 278 K, for $\alpha/\epsilon = 1$, to 494 K, and a sphere plated with magnesium oxide white paint can drop to 156 K. Single-sided flats can rise from the nominal 393 K, for $\alpha/\epsilon = 1$, to about 700 K!

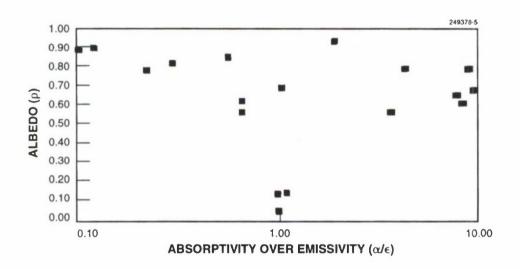


Figure 5. Reflectivity vs. $\alpha \in for spacecraft materials$.

There is a paucity of data on the dependence of ϵ with variables such as wavelength and temperature. Ross et al. [13] report measurements on silicon solar cells where they give the value of $\alpha = 0.78$, and emissivity changes from 0.68 at -50°C to 0.78 at 150°C (see Figure 6). Blair et al. [14] report on extensive measurements on a number of materials with emphasis on solar cells manufactured by AEROJET. They measure the emissivity dependence on reflectance angle, temperature, and wavelength ranging from 2 to 24 μ (see Figure 7). From 2 to 18 μ the emissivity is generally between 0.8 and 0.9. There is a decreasing trend in emissivity for wavelengths greater 18 μ , although the data are inconclusive.

Dow inverted this table. Dow used 1400 W/m² for the solar power, whereas 1360 W/m² is used in this analysis.

They report emissivity of about 0.9 at 200 K and less at 373 K. This is the opposite trend from that reported by Ross et al. Dow reports that the emissivity of white paint changes from 0.9 at 50 μ to 0.2 at 75 μ and to 0.1 at 100 μ . Therefore, some dependence of emissivity on temperature is expected, although even the sign of the slope is unknown. More probably we would expect the emissivity to decrease with increasing wavelength. This is consistent with Drudes's theoretical relation between the resistivity ϱ of a metal for a definite wavelength [15].

$$\epsilon_{\lambda} = 0.365 \sqrt{\frac{Q}{\lambda}}$$
.

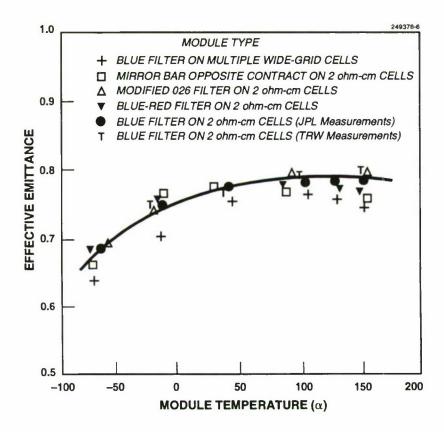


Figure 6. Emissivity vs. temperature for silicon solar cells.

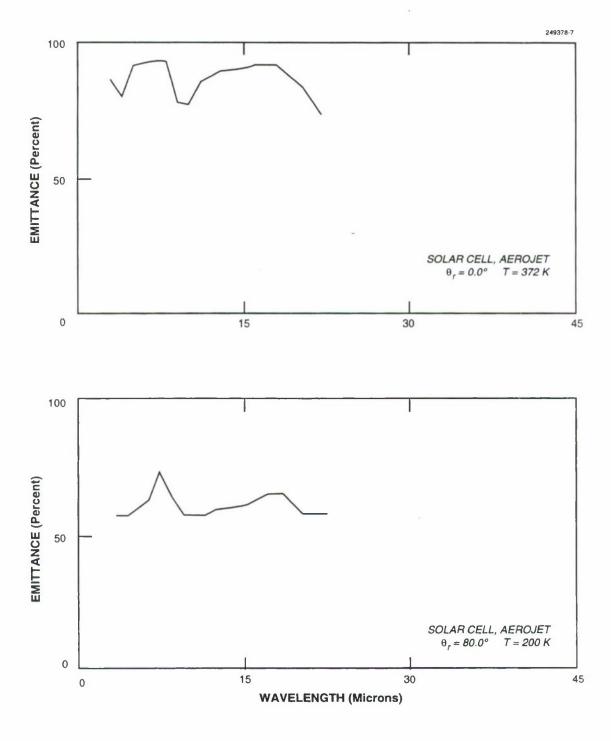


Figure 7. Emissivity vs. wavelength for silicon solar cells.

Currently, it is not generally known if the values of α/ϵ change with time in space. For optical solar radiators, α/ϵ does change from 0.08 to 0.2 or more over 10 years [18]. This is inferred from the increase in electronic component temperature. Results from the Long Duration Experiment Facility (LDEF), in orbit for almost six years, provide some indication. As summarized by Dow, for anodized aluminum (the primary LDEF surface coating), the values of ϵ did not change significantly, whereas the value of α increased by as much as 16%, depending on the location of the sample. However, white paint exhibited increases as large as 100%. Such phenomenon may help explain the high and low temperatures reported here.

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TABLE 2
Typical Absorptivity and Emissivity for Spacecraft Materials

Material	ρ	α = 1-ρ	E	α/ε	T1	T2
AIO ₃ on Buffed Aluminum	0.87	0.13	0.23	0.57	243	289
Vapor Dep. Gold on Glass	0.96	0.04	0.02	2.00	333	396
Vapor Dep. Silver on Glass	0.81	0.19	0.02	9.50	492	585
Solar Cell (IUE)	0.14	0.86	0.84	1.02	282	335
Stainless Steel (Polished)	0.58	0.42	0.11	3.82	392	465
Fiberglass	0.15	0.85	0.75	1.13	289	344
Polished Aluminum 6061	0.81	0.19	0.04	4.52	408	485
Unpolished Aluminum 6061	0.63	0.37	0.04	8.81	483	574
Anodized Aluminum	0.58	0.42	0.63	0.67	254	302
Silver	0.96	0.04	0.02	2.00	333	396
Gold (Plated)	0.70	0.30	0.03	10.0	498	593
TiO White Paint	0.80	0.20	0.90	0.22	192	228
Black Paint(#M Velvet)	0.05	0.95	0.92	1.03	282	336
Aluminum Paint	0.71	0.29	0.27	1.07	297	353
1.0 mil Aluminized Mylar*	0.84	0.16	0.54	0.30	207	247
1.0 mil Silverized Teflon†	0.92	0.08	0.66	0.12	165	196
1.0 mil Aluminized Kapton‡	0.64	0.36	0.54	0.67	254	302
Magnesium Oxide White Paint	0.91	0.09	0.90	0.10	158	187
Platinum Foil	0.67	0.33	_0.04	8.25	475	565
* Trade name for polyethylene tereph	nthalate					
† Trade name for fluorinated ethylene	e propylene					
t Trade name for polyamide						

[‡] Trade name for polyamide

5. CALIBRATION ISSUES

The IRAS data received from Groningen was processed with the calibration described in Beichman's report [9]. Since that publication, a number of analyses have questioned the calibration [16,17,18,19,20,21]. Some consider the calibration confirmed (Aumann [22]), whereas a number of analyses suggest the various wavebands are in error by 2% to 12%. The proposed corrections are listed in Table 3. The weight of evidence suggests that some calibration correction may be warranted. The calibration correction adopted for this analysis is also given in Table 3. In this analysis, there is no way to quantify or assess the absolute calibration, as we do not have independent information about the satellite emissivity, absorptivity, or temperature. It is the determination of these quantities that is the objective of this analysis.

TABLE 3
IRAS Calibration Errors

Source	12 μ	25 μ	60 µ	100 μ
Aumann	0	0	0	0
Tedesco	0	10%	10%	?
Gillett	0	6%	0	?
Kirby et al.	0	10%	0	0
Cohen et al.	2%	6%	3%	12%
Used Here	0	0	0	NA

6. DATA PROCESSING

Each detection report contains, among other information, the epoch, direction in right ascension and declination, angular rate crossing the focal plane, and the flux, in uncolor-corrected Janskys, for each detector registering a "hit."

The objectives of this analysis are to identify those detections of known satellites and to analyze the radiometric data for information about IR observations of artificial earth satellites. There were four steps in preparing the data for analysis.

- 1. The observed direction of all detections in the Groningen debris data base was compared with directions computed from an ephemeris of the IRAS satellite and the catalogue of known satellite element sets. Observations were accepted if the observed and computed direction agreed to 0.60° or better and the observed and computed angular velocity agreed to 0.5 arcmin/sec.
- 2. The individual hits for each detection were screened. The intention was to eliminate hits where the satellite image (assumed a point source) did not completely cross a detector, as a partial hit would give a biased flux estimate. The screening was done as follows. As seen from the focal plane geometry (see Figure 2), the detectors for each waveband are in two rows with some overlap. Therefore, in each detection one would expect that two hits would be registered for each waveband. In cases where three or more hits are registered, it is assumed that some of them are at a detector edge and therefore do not give a good flux measurement. Therefore, the screening was that for detections with three or more hits, the two hits with the largest flux density measurement were accepted. This eliminated some, but probably not all, partial detector hits. An example of the editing of the data for SBS1 (12065) is presented in Table 4. The sighting time for each detection is given as the day of the year 1983. The record number is a unique identifier in the Groningen IRAS debris data base. The object number is 12065. The observed in-scan velocity is given in arcmin/sec. In each detection, all the hits are given, and the detector is identified. A number of triple hits are observed. The discarded hit is noted with an asterisk. Hits were discarded for other reasons. The detection on day 222.032 was discarded because of the anomalous in-scan velocity. Other hits discarded, based on a statistical test, were in detections on days 71.535 and 72.537.
 - 3. The detections for each satellite were collated.
- 4. The flux density measurements were color corrected and analyzed to obtain physical properties of the satellite using other observed information such as the range and physical size. Information such as temperature and emissivity were derived.

TABLE 4
Detector Hits for SBS1

Sighting Time	Rec	(Obj	InsV	D12	Flux12	D25	Flux25	09Q	Flux60
48.55410004	2004	12065	4.38	8 82	1.83	42	2.22	34	1.39
142.00761414	55033	12065	4.64	25 83 52	1.70 0.63* 1.90	44	2.43	13	0.43
69.53096771	79019	12065	4.46	51 27	1.86	39 19	2.80	अ अ	0.91
70.53330231	79814	12065	4.46	8 8	1.98	43 40 16	0.65° 2.59 2.32	32 13 8	1.09 0.65* 1.23
71.53556824	80687	12065	4.47	24	1.76	44	0.42*	33	0.85
72.53787231	81955	12065	4.47	8 8	3.37*	45	2.13	37	1.22
73.54016876	82421	12065	4.53	22 SS	1.65	8 23	2.49	38 15	2.19 1.39 0.62*
222.03245544	146290	12065	3.42	198	0.56*	9	2.26*		

7. GENERAL RESULTS

Table 5 summarizes the result of correlating with the satellite catalogue. The number of correlations is given for a variety of error bounds. The 2072 detections with angle errors less than 0.6° are believed to be valid satellite detections. The remaining detections contain uncatalogued objects (UCTs), debris, and false detections. These will not be discussed further.

TABLE 5
IRAS Data Correlation Results

Number of IRAS sig	htings compared with	catalogue	136304
Number of correlation	ons		
Crude filtering			30600
< 4°	•		9955
< 1°			2469
< 0.6°			2072
Number of satellites	correlated		465
Statistics	vs. IRAS DB	vs. Corr	relations
Correlations	22%		
< 4°	7%	33%	
<1'	2%	8%	
< 0.6°	2%	7%	

The Wesselius's original publication [8] displayed the data in an interesting way. From the observed flux ratio, say $f_{12\mu}/f_{25\mu}$, one can determine the temperature of the body, assuming the emissivity is constant. One would expect the temperature measured with the $f_{12\mu}/f_{25\mu}$ ratio to be the same as that obtained from the $f_{25\mu}/f_{60\mu}$ ratio. This is plotted in Figure 8 from Wesselius et al. for the 1500 detections with fluxes greater than 3 Jy. The wide scatter belies our expectation. The same plot is provided for the correlated data before the screening for partial hits (see Figure 9). Instead of selecting detections based on large flux density, we have selected detections based on known satellite detections. A similar scatter is evident. Next, the same plot is provided for temperature derived from the screened flux density measurements (see Figure 10). In this case, the values fall on a straight line, and the scatter is significantly

reduced. From the $f_{12\mu}/f_{25\mu}$ flux ratio, the average temperature of about 300 K is clearly evident. We believe the large scatter in the original IRAS data is due in part to partial detections. In principle a partial detection could be corrected given precise knowledge of the IRAS position and pointing and the satellite position.

Another way to characterize the IR satellite measurements is given in Figure 11. Here, the $12-\mu$ flux density is presented as a function of the range to the observed target. The envelope corresponds to the $1/r^2$ dependence, and its position depends on the sensitivity of IRAS and the maximum size of objects in 1983. A number of detections are at a greater-than-geosynchronous range. These are genuine since they are Astron, Exosat, and other known satellites. The individual fluxes are given in Appendix A.

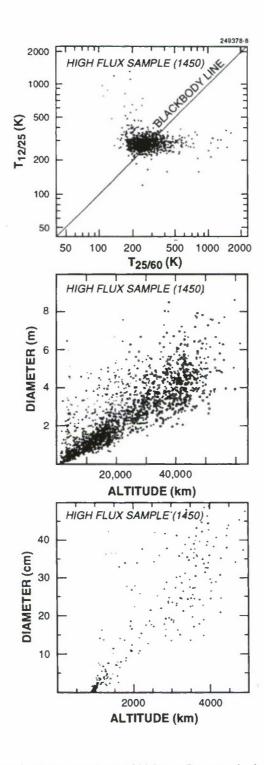


Figure 8. Temperature from 1500 large flux sample detections.

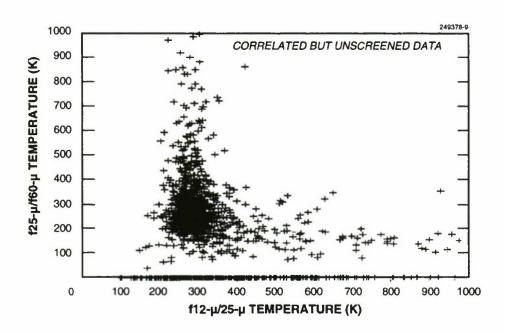


Figure 9. Temperature from correlated and unscreened data.

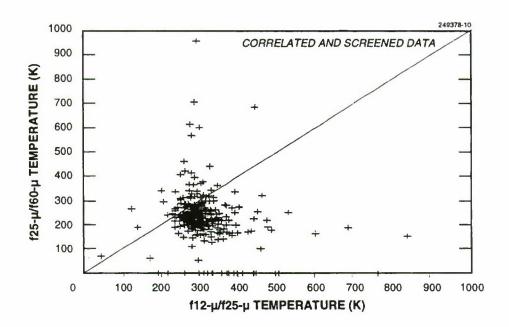


Figure 10. Temperature from correlated and screened data.

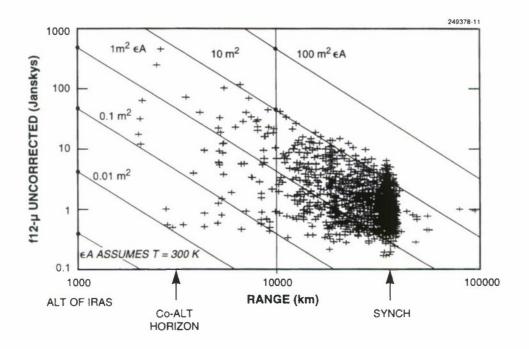


Figure 11. Flux density vs. range.

8. REFERENCE SPHERES

Isothermal spheres are the simplest and easiest satellite shapes to analyze. A single IRAS focal plane crossing has been correlated with the polished aluminum Lincoln Radar Calibration Sphere LCS1 (SSC # 1361). Although there is only one detection, the analysis of this data is instructive. The temperature is determined to be 480 K using the screened uncolor-corrected flux density ratio of $J_q^{12\mu}$ = 17.69 Jy at 12 μ over $J_q^{25\mu}$ = 11.22 Jy at 25 μ . The radiant flux density of LCS1 at 480 K is $\mathscr{F}_{12\mu}(480)$ = 134.8 W/ μ /(effective ϵA) at a 12- μ reference wavelength.

The IRAS color correction (Beichmann, Table VI-26) adjusts the reported 12- μ flux density to $J_{obs}^{12\mu} = 16.55$ Jy, which is equal to $F_{obs}^{12\mu} = F_q^{12\mu}/K_{12\mu} = 3.45 \times 10^{-13}$ W/(m² μ) spectral irradiance. Now at a range of 2001 km, determined from the a posteriori ephemeris, the observed spectral radiant intensity is $\pi r^2 F_{obs}^{12\mu} = 4.34$ W/ μ .

Now the effective area ϵA is the observed spectral radiance at 12 μ (W/ μ) divided by the Planck function value of the radiant flux density [W/(m² μ)] at the same reference wavelength.

$$\epsilon A = \frac{\pi r^2 F_{obs}^{12\mu}}{\mathscr{F}_{12\mu}(480)} = 0.0322 \ (m^2)$$

Thus, ϵA is 0.0322 m², but since the projected area for LCS1 (1361), $A = 1 \text{ m}^2$, the emissivity is $\epsilon = 0.0322$.

The ratio of solar absorptivity to emissivity $(\alpha/\epsilon)^{1/4}$ is equal to the observed temperature, 480 K, divided by 278 K, the characteristic free-space temperature for an isothermal sphere. Thus, $\alpha/\epsilon = 8.89$, $\alpha = 0.286$, and the reflectivity is $\rho = 1 - \alpha = 0.7139$.

Data on the optical properties of various spacecraft materials give values for aluminum as shown in Table 6.

The radar calibration sphere may have a less polished surface than when launched a quarter century ago, thereby increasing the absorptivity—i.e., reducing the reflectivity—toward the values given for unpolished aluminum.

The IRAS result can be compared with visible band measurements. The apparent visual brightness resulting from a specular sphere with $\rho A = 0.714 \text{ m}^2$ at a 2000-km range is given by

$$M_v = -26.78 - 2.5 \log \left(\frac{0.714}{4\pi (2x10^6)^2} \right) = 7.84 M_v$$

TABLE 6
Optical Properties of Aluminum

	α	E	α/ε	ρ	T (K)
LCS1 Observed	0.286	0.0322	8.89	0.714	480
Polished Al	0.19	0.042	4.52	0.81	408
Unpolished Al	0.63	0.042	8.81	0.63	483

This calculated brightness can be compared with the brightness observed by W. Beavers [2] as a function of phase angle. The value derived from IRAS for M_{ν} is 7.84 M_{ν} at 2001 km. This can be referred to geosynchronous distance, $\approx 36,000$ km, with an adjustment of 6.275 M_{ν} . Therefore, IRAS predicts 14.11 M_{ν} at geosynchronous range. This is in moderate agreement with Beavers's observations reported at geosynchronous range that give a value, at 90° phase angle, of 14.0 M_{ν} . Beavers notes that his observations indicate that 12% of the energy is diffuse. Additional LWIR and visible band measurements at the same epoch are required for a high-accuracy comparison.

Finally, consider the thermal response of LCS1, a 1-m sphere, with a 2-cm-diameter solid aluminum sphere covered with Martin Black with $\alpha = \epsilon = 0.999$ as it enters the earthshadow. The cooling rate, neglecting earth-upwelling radiation, is

$$\frac{dT}{dt} = -\frac{5.67 \times 10^{-8} T^4 \in A (1-\xi)}{c_p M} \left(\frac{kW}{J}\right) ,$$

where c_p is the specific heat, M is the mass, and $(1 - \xi)$ is the fraction of the sphere radiating to free space. Now, the temperature of a sphere is written as

$$T=278\left(\frac{\alpha}{\epsilon}\right)^{\frac{1}{4}}$$
,

giving

$$\frac{dT}{dt} = -\frac{338.65 \alpha A (1-\xi)}{c_p M} \quad .$$

Now, the area-to-mass ratio of the two spheres, $(A/M)_{(1361)} = 0.0285 \text{ m}^2/\text{Kg}$, and $(A/M)_{(2 \text{ cm})} = 0.0252 \text{ m}^2/\text{Kg}$, is virtually the same. The cooling rate for LCS1 (1361) is therefore expected to be about 0.286 that of the 2-cm emissive calibration sphere coated with Martin Black, which results in a temperature excursion less than 10 K. Therefore, LCS1 would provide a relative stable temperature reference.

9. SPIN-STABILIZED CYLINDERS

Spin-stabilized, solar-cell-covered cylinders appear to be simple objects for radiometric analysis. There were 80 such nonmilitary satellites in orbit during the IRAS sky survey compared to 20 three-axis stabilized satellites. The putative simplicity of spin-stabilized cylinders led to early analysis of that data.

The first geostationary satellite, Syncom B, a 28-in-diameter and 15-in-high spun cylinder, was launched in 1963. The cylinder weighed 85 lb. The solar power was 30 W, and the satellite had a lifetime of 18 months. Now, 30 years later, Hughes (the same company that built Syncom) deploys two-ton cylinders, 12 ft in diameter and 30 ft high with 2000 W of solar power at the end of a ten-year life.

The first cylinders employed dipole antennas, sometimes in phase switched arrays, to provide moderate gain in the earth direction. Later, satellites used high-gain parabolic reflectors on a despun platform at the top of the cylinder. The other end of the cylinder was left open, and a second concentric solar-cell-covered cylinder slid down over the apogee kick motor after geosynchronous orbit was achieved. These geosynchronous orbits have nearly 0° inclination and circular orbits with the spin axis maintained normal to the orbit plane, i.e., normal to the earth's equatorial plane. The IRAS orbit geometry results in observations of geosynchronous satellites near quadrature, i.e., at an illumination phase angle between 80° and 100° . The measured flux density from these cylinders is expected to be a weak function of phase angle, which is in contrast to flat solar panels that exhibit a $\cos(\phi)$ relationship. The LWIR detections of spin-stabilized cylinders are therefore expected to be the simplest to understand.

The simple LWIR model used here treats only the main cylindrical body. The LWIR radiance of the antenna and despun platform are ignored for two reasons. First, they are covered with reflective thermal insulating material. Second, because of the IRAS geometry, the observations are made when the parabolic antenna is edge-on to the sun, it is cool, and its radiation is ignorable. The LWIR energy radiated from the open bottom of the cylinder is also ignored in the energy balance calculation. Now, the cylinder, spinning on the order of once per second, is certainly near thermal equilibrium, and only the projected area is considered. Incidentally, the cylindrical geometry produces only $1/\pi$ as much power as an equal number of solar cells on a flat panel normal to the sun.

From information about solar cell absorptivity and emissivity it is known that about 10%-15% of the incident solar power is reflected by the solar cells $\rho = 1 - \alpha$. About 15% of the incident solar power is converted to electrical power. This leaves about 75% of the solar power to be reradiated to achieve thermal equilibrium. The free-space temperature of a solar panel is changed by

$$(\alpha/\in)^{1/4}$$

where $\epsilon \approx 0.85$, and $\alpha \approx 0.75$. Note that $\alpha = 1 - \rho - \eta$, where the symbol for reflectivity is ρ and the symbol for solar cell efficiency is η . It is the fourth root of the ratio of this net absorptivity, $\alpha \approx 0.75$, to emissivity, $\epsilon \approx 0.85$, that determines the temperature of a given-shape solar panel.

Most cylindrical payloads were built by Hughes, including the giant HS-393 class, which includes SBS-6 and INTELSAT-6 (all launched after 1983). The Hughes Galaxy class, HS-376, Figure 12, was

selected for initial analysis because of the number of IRAS detections, ≈ 60, on eight individual satellites that were obtained. In addition, they had flux densities exceeding 2 Jy. Now, the HS-376 class was the first to radiate heat from the electronics using a cylindrical optical solar radiator (OSR). Previous designs radiated the heat from the electronics upward toward the antenna via the despun platform. To obtain the simple model for the cylinder, it is assumed that the radiator power is equal to the reradiated power of the solar panel it replaced. The HS-376 was also the first to increase the solar cell collection area by sliding a concentric solar cylinder down to cover the apogee kick motor after geostationary orbit was achieved.

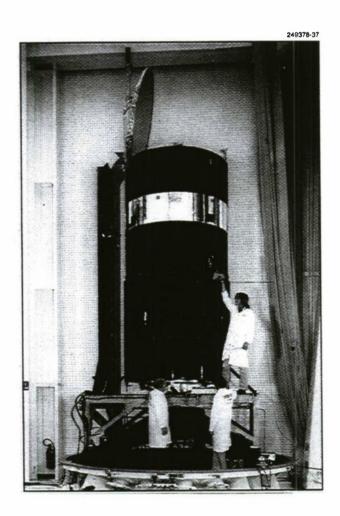


Figure 12. Hughes HS-376 satellite.

The eight Hughes cylinders observed by IRAS are listed in Table 7. Figure 13(a), (b), and (c) plots the detections as a function of time; there is no indication of a temporal trend, and we would expect none. This confirms the constancy of the IRAS calibration. Therefore, we have taken an average of all the screened detections and obtained the mean values given in Table 7, which are accompanied by the formal uncertainty. In general the average flux is obtained to \pm 0.1 Jy. With this statistical population, the accuracy of a single hit is estimated as \pm 0.6 Jy. The object temperature is obtained as described above. We adopted the temperature derived from the $f_{12\mu}/f_{25\mu}$ ratio as the object temperature for further analysis. Given the temperature, one can then recover physical properties of the satellite as follows. The observed flux densities are color corrected to find the true flux density. Since the range is known from the ephemeris used for correlation and the self-emitting projected surface area of these satellites, $A_p = 8.84 \text{ m}^2$, the emissivity can be calculated in each waveband from

$$\epsilon_{\lambda_o} = \frac{J_q^{\lambda_o}}{K_{\lambda_o}(T)} \frac{\pi r^2}{A_p J_o^{\lambda_o}(T)} .$$

The calculation of emissivity for the eight Hughes cylinders is given in Table 8. Note that the 12-and $25-\mu$ band emissivities are in good agreement with each other, as they should be since they were assumed to be so for calculation of the temperature. The 12- and $25-\mu$ emissivities are also in good agreement with published values (Table 2), whereas the $60-\mu$ values are not. We assume that either there is an (unlikely) calibration error in the $60-\mu$ band or there is a real difference in solar cell emissivity at $60~\mu$. Finally, the solar absorptivity of the satellites can be calculated as described above. The absorptivity calculated in this way is also listed in Table 8.

From the HS-376 results, a convenient rule of thumb is postulated relating the observed $12-\mu$ flux $J_{12\mu}$ to projected solar cell area A_p

$$A_p \approx 5 J_{12\mu} \qquad (m^2) .$$

This leads to a relation between power and observed flux

$$W \approx 500 J_{12 \mu} \qquad (W)$$

for nominal 10% solar cell efficiency.

TABLE 7 IRAS Detections of Hughes HS-376 Cylinders

Sid	Name	12 μ		25 μ		ф09		T12/25	T25/60
		Jansky	#	Jansky	*	Jansky	*	¥	×
12065	SBS-1	1.86±0.04	13	2.42±0.11	12	1.20±0.19	14	297	223
13069	WESTAR 4	1.59±0.06	16	2.30±0.08	16	1.01±0.12	16	284	247
13269	WESTAR 5	1.78±0.03	9	2.48±0.22	5	1.02±0.19	5	288	263
13651	SBS-3	2.03±0.10	2	2.68±0.10	2	1.25±0.10	2	296	235
13652	ANIK C3	1.87±0.10	12	2.67±0.11	12	1.00±0.34	10	286	287
13431	ANIK D1	1.83±0.11	10	2.74±0.08	10	1.14±0.08	-01	279	261
14158	Galaxy 1	1.61±0.19	12	2.39±0.16	12	1.04±0.39	12	280	250
14134	PALAPAB1	1.52±0.10	2	2.64	-	0.98±0.10	0	260	291

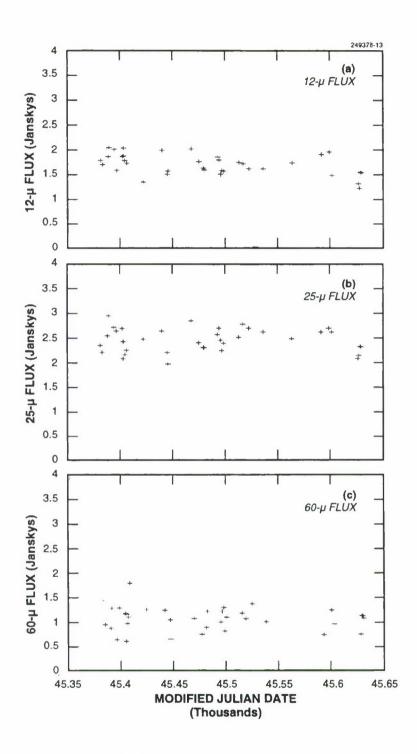


Figure 13. Flux vs. time for Hughes HS-376 cylinders.

The general result for analysis of spin-stabilized cylinders is that relatively simple models can give important information about these satellites. Assuming that solar cells are the primary source of LWIR flux seems a workable assumption. It allows inference of effective emissivities and possibly some inference about the solar cell power available.

The HS-376 spacecraft observed by IRAS were assumed initially to be more or less identical and were expected to yield the same flux densities and temperatures. The values of flux density for a given spacecraft were averaged without adjustment for possible seasonal variations. However, individual spacecraft had important differences, some of which could explain the difference in observed characteristics. WESTAR 4 and 5 have shorter solar panels and corresponding lower power [23]. If the radiator is the same size, which is not documented, the two WESTARs have only 7.8 m² projected area compared with the other HS-376 with 8.84 m². Thus, the WESTARs have 0.88 of the area, and this is in good agreement with the 12- μ flux density ratio of 0.87 of the average of the two WESTARs compared with the two ANIKS (1.613 Jy/1.851 Jy). In addition, the two SBS satellites had IRAS-derived temperatures about 15 K higher than the other spacecraft. This may be due to the smaller number of 20-W TWT amplifiers employed. Yenne [24] reports that SBS satellites had only 10 TWTs compared with the usual 20 on other satellites. Another possible source of difference is that early spacecraft were launched without baking out the solar cell panels to remove the water from the hydroscopic solar cell adhesives. Both SBS-1 and WESTAR 4 were not baked out before launch, resulting in a few percent lower solar power for several months during outgassing in orbit. However, outgassing should have been completed long before the IRAS data was taken.

The most startling difference among spacecraft is the switch from a K7 Spectrolab solar cell to the mixture of K7 and K4 $\frac{3}{4}$ cells on the Galaxy 1. The K7 cells have been featured as having the highest peak solar output (19 mW/cm²), albeit with higher temperature resulting from a higher absorptivity, $\alpha \approx 0.84$. The K7 cells are thin (2.5 mils) and have a textured front surface together with an aluminized rear coating that reflects near-IR back through the cell for increased power generation. The K7 was particularly extolled for use on the deployable solar panel where the characteristic excess heat could be easily radiated [25]. Apparently, the K7 cells on the fixed panels flanking the radiator made it more difficult to control the temperature of the electronics and batteries behind the radiator panel.

The design of high-power, long-lived, solar-cell-covered cylinders is strongly driven by the temperature control required for electronic components and batteries, which power the RSO during earth eclipse near equinoxes. The solar cells are cemented to 1/2-in-thick, Kevlar-faced, aluminum honeycomb structural panels. In the early Hughes 376 series, the new K7 high-efficiency solar cells were used on both the fixed upper (forward) and deployable (lower) cylinders. The switch to the cooler K4 ¾ solar cells suggests a temperature control issue.

TABLE 8 Emissivity and Absorptivity for HS-376 Cylinders

Sid	Name	J12/J25	T(K)	J12/J25*	E _{12u}	6 _{25µ}	109 ₃	۵
12065	SBS-11	0.7690	297	0.9645	0.73	0.72	0.98	0.75
13069	WESTAR 4	0.6910	281	0.8749	0.85	0.88	1.02	0.74
13269	WESTAR 5	0.7160	288	0.9055	0.92	0.91	66.0	0.82
13651	SBS-3	0.7570	295	0.9519	0.82	0.82	1.03	0.82
13652	ANIK C3	0.7018	284	0.8882	0.89	0.89	0.82	0.77
13431	ANIK D1	0.6690	277	0.8480	0.97	0.97	1.03	0.76
14158	Galaxy 1	0.6763	278	0.5570	0.83	0.83	0.93	0.67
14134	PALAPAB1	0.5739	258	0.7307	1.13	1.13	1.00	0.67
* Color Corrected See Table 7 for pl Free-space tempe	nysic eratu	al dimensions. re = 295 K.					:	

The OSR has quartz panel sections with second surface mirrors and black backing coatings. The initial solar absorptivity-to-reflectivity ratio is less than 0.08. Thus, the radiator cooling section, between the two upper solar panels, acts as a one-way mirror that reflects sunlight and allows heat from the TWTs, nickel cadmium battery cells, etc., to radiate into free space. The electronics shelf, including the RF output TWTs behind the radiator, is on the despun platform with the antenna and feeds. The batteries spin with the solar panels. The rise of spacecraft temperature of several degrees over its lifetime has been attributed to the slow degradation of the OSR's α/ϵ .

The power system for spacecraft usually comprises two redundant main array buses together with battery charge and trickle arrays. The nearly 14,000 solar cells of the HS-376 spacecraft are connected in various "strings" of series-parallel configurations, with protective diodes, to allow graceful degradation in the event of string failure.

Hughes's modeling of solar cell power, voltage, and current predicts performance within a few percent. This is confirmed by several years of telemetry. Temperature is also modeled, but telemetric data agrees with models to only $\pm 6\%$ of the typical solar panel temperature (283 K). The seasonal variation of bus voltage due to the 23.5° obliquity of the ecliptic and the earth orbit eccentricity is clearly observed in the models. The temperature cycle is not so apparent since at eclipse season, when more solar power is available during sunlight, more power must be used to recharge the batteries following the eclipse.

More detail about solar cell panels is given for the HS-376 successor, the HS-393 [26]. Table 9, taken from Fodor et al. converted to degrees Kelvin, summarizes the predicted array panel temperatures by season and for the beginning and end of life. $K4\frac{3}{4}$ cells are used only on the fixed panels, and the K7 cells are used on the deployable panel. Though the total power of the two buses is over 2000 W, the number of solar cells required is not larger than for HS-376 because much larger solar cells have been developed to simplify fabrication. Note that the temperature of the $K4\frac{3}{4}$ cell panel is higher by 5–9 K than the K7 panels, except near winter solstice. This indicates that equipment temperatures are better controlled with the $K4\frac{3}{4}$ cells near the radiator. Incidentally, the lower panel falls to 191 K before emerging from eclipse. Also note the trickle charge panel rises to 306 K at winter solstice as the sunlight partially illuminates the inside of the lower cylinder.

It was recently learned that in the mid-1970s solar cell improvements foretold an operational increase in watts per square meter of 30% to 50%. Allison et al. [27] describes an improved "violet" cell (announced in 1972) with vastly improved response to the blue-violet solar spectrum that achieved at least a 30% increase in solar power over the conventional cell in use. The COMSAT nonreflective (CNR) cell announced in 1974 offered a 50% increase in power above the conventional cell as well as increased radiation hardness. The solar efficiency η is 15.4%.

TABLE 9
Nominal Solar Panel Temperature for HS-393 Satellites

		K4¾		K7 Deployable	
Date	Years in		Array Temper	ature (Deg K)	
Date	Orbit	Panel A1	Panels C1-C4 MC1	Panel MC2	Panel Tc
Summer Solstice	0	275.8	266.9 266.9	268.0	265.2
Autumnal Equinox	0	278.0 280.8	275.8	268.0 277.4	265.2 272.5
Winter	10 0	283.0 276.3	275.8 297.4	277.4 289.1	272.5 305.8
Solstice	10	278.6	298	289.7	305.8
21 March Post-Ecl	0	191.3	144.7		

Meulenberg [20] also advises that $(\alpha - \eta)/\epsilon$ is a better indicator of cell temperature than α/ϵ . Table 10, taken from the COMSAT paper, tabulates the properties of conventional cells, the violet cell, and the CNR cell. We have added the calculated cell temperature modeled on a cylindrical satellite

$$T \doteq 295 \left(\frac{\alpha - \eta}{\varepsilon}\right)^{\frac{1}{4}}$$
.

The CNR cell clearly has a higher temperature, and the paper suggests that a specific cover slide would increase the emissivity from 0.803 to 0.843, thereby somewhat ameliorating the temperature increase. The K7 cells used on the early Hughes HS-376 class satellites are more or less equivalent to the CNR cells. IRAS data on seven of the eight observed satellites were reexamined—the PALAPA B1 detection was excluded—to correlate the observed temperatures with the solar cell models. The results are given in Table 11.

TABLE 10
Solar Cell Types and Temperature

Туре	Electrical Efficiency	Solar Absorptivity α	Normal Emittance	(α-η)/ε	Temperature
Conventional (Pre-1976)	0.101	0.725	0.803	0.777	277
(F16-1970)	0.101	0.725	0.843*	0.74	273.6
"Violet"	0.14	0.81	0.803	0.834	281.9
	0.14	0.81	0.843*	0.795	278.5
COMSAT Non-	0.155	0.906	0.803	0.935	290.1
reflective (CNR)	0.155	0.906	0.843*	0.891	286.6

IRAS data on the ANIKs and the WESTARs show good agreement with the model temperature. Satellite Business System Consortium of COMSAT, SBS-1 and -3 were reported to have only 10, instead of 20, 20-W RF power TWTs [24]. With the assumption that only about one-half the bus power was used, and the remaining power was reradiated at LWIR instead of RF, the CNR cell equivalent of η was one-half of 15%. The result is in good agreement with the observed temperature and the solar cell model, thus explaining the approximate 9° difference between the 295 K for the SBSs and the 286 K for the ANIKs and WESTARs.

The Galaxy 1 with $K4\frac{3}{4}$ cells on the fixed upper cylinder and K7 on the deployable panel was modeled by averaging the properties of the old solar cell with the CNR cell properties from Table 10. The resulting model is within about 6 K of the IRAS measured temperature.

The good agreement found between the simple model and the IRAS measurements should not lead to the conclusion that the simple model is an adequate treatment of a very complex thermodynamic system. Rather, the neglected details, such as the antenna thermal flux and radiation from the open deployed cylinder, tend to compensate for one another.

TABLE 11
Physical Properties for HS-376 Cylinders

SSN	Satellite	Model Temp	IRAS Temp	Laboratory ©	IRAS €	Laboratory α-η	IRAS (α-η)*
13431	ANIK D1	286.6	277.1	0.843	0.973	0.751	0.763
13652	ANIK C3	286.6	283.6	0.843	0.892	0.751	0.768
13069	WESTAR 4	286.6	281.4	0.843	0.891	0.751	0.744
13269	WESTAR 5	286.6	286.4	0.843	0.917	0.751	0.822
12065	SBS-1	293.7	296.9	0.843	0.728	0.828	0.747
13651	SBS-3	293.7	294.5	0.843	0.818	0.828	0.815
14158	Galaxy 1	280.1	278.5	0.843	0.836	0.688	0.670
* (α-η)	is tabulated as	α in Table	9 IRAS me	asurement.			

The previous analysis concentrated on IRAS detections of the eight HS-376 spin-stabilized cylinders. The moderate consistency of flux densities and temperatures encouraged analysis of the remaining observations of spin-stabilized cylinders. Table 12 gives some physical information about many of the satellites in orbit during IRAS observations as well as the spin-stabilized cylinders observed: size, weight, electrical power, type, name, and manufacturer. Table 13 gives the observed flux densities—uncorrected for color temperature—and the calculated temperatures, emissivities, and absorptivities. All other observed cylinders had smaller projected areas than the HS-376 class, and almost all the reported flux densities were smaller than those for the HS-376s. Two INTELSAT 4s (4881 and 6052) and three INTELSAT 4As (8620, 10557, and 10778) were observed to have temperatures between 291 and 298 K, with a mean near 295 K. Both classes, HS-312 and HS-381, have approximately the same projected area.

TABLE 12 Payloads Detected by IRAS

Name	TELSTAR 2 (A-41)	LCS 1	ATS 1	ATS 3	LES-6	NATO 1	INTELSAT 3 F-8	INTELSAT 4 P-2	NATO 2	INTRESAT 4 P-5	ANIK A1 (TELESAT-1)	ANIK A2 (TELESAT-2)	WESTAR 1	ANIK A3 (TELESAT-3)	GOES 1 (SMS-C)	INTELSAT 4A P-2	LAGEOS I	COMSTAR 1	MARISAT 2	PALAPA 1	COMSTAR 2	MARISAT 3	KIKU 2 (ETS-2)	GOES 2	HIMAMARI (GMS-1)	INTRESAT 4A F-3	INTELSAT 4A P-6	ANIK BI (TELESAT-4)	AYAMB 2 (ECS-2)	GORS 4
Manufacturer							TRW	Rughes 312		Hughes 312	Hughes	Hughes	Hughes 333	Hughes	Philco Ford	Hughes 353		Hughes 351		Hughes	Hughes 351	Hughes 356	NASDA	Philco Ford	Hughes	Hughes 353	Hughes 353			Hughes 371
Remarks			Gravity Stabilized	Gravity Stabilized				2.45-m antenna		2.45-m antenna			1.5-m entenna			2.45-m antenna		3.38-m entenna		1.5-m antenna	3.38-m antenna	1.65-m antenna				2.45-m antenna	2.45-m antenna		lost contact on injection	
TV	.581	1.003	2.059	2.599	2.044	1.110	1.477	6.740	1.110	6.740	2.782	2.782	2.040	2.782	2.560	6.740	.283	6.897		3.610	6.897	3.450	1.148	4.370	3.350	6.740	6.740	4.644	1.330	3.350
	•			14										• •	~	•														
Diameter (m)	98.	1.13	1.42	1.42	1.22	1.37	1.42	2.39	1.37	2.39	1.83	1.83	1.90	1.83	1.90 2	2.39	09.	2.44		1.90	2.44	1.90	1.40	1.90	2.15	2.39	2.39	2.17	56.	2.16
		1.13									1.52 1.83		1.90	1.03	1.90	2.39		2.82 2.44			2.44					2.82 2.39			1.40 .95	
Diameter (m)		1.13		1.42	1.22				10.			1.83	1.60 1.90	1.52 1.03	1.90	2.39				1.90	2.82 2.44						2.82			2.16
Height (m) Diameter (m)				1.42	1.66 1.22	10.	1.04	2.82	.01	2.82	1.52	1.52 1.83	300 1.60 1.90	1.52 1.03	1.90	2.62 2.39				1.90 1.90	2.82 2.44	2.42		2.30	3.45	2.82	600 2.82			2.16

TABLE 12 (Continued)
Payloads Detected by IRAS

814	Mass (kg)	Power (watts)	Height (r	Ē	Diameter (m)	77	Remarks	Manufacturer	Маже
1 1 1			 	-					
12065		1100	4.78		2.16	8.600	1.82-m antenna	Hughes 376	SBS 1
12295			2.80		2.10	5.880		NASDA	KIKU 3 (ETS-4)
12309			2.82		2.44	6.897	3.38-m antenna	Hughes 351	COMSTAR 4
12677		225	3.45		2.15	3.350		Hughes	HIMAMARI 2 (GMS-2)
13069		1000	4.78		2.16	7.80	7.800 1.82-m antenna	Hughes 376	WESTAR 4
13269		1000	4.78		2.16	7.800	1.82-m antenna	Hughes 376	WESTAR 5
13431		006	4.78		2.18	0.600	1.65-m antenna	Hughes 376	ANIK DI (TELESAT-6)
13651		1100	4.78		2.16	8.600	1.82-m antenna	Hughes 376	SBS 3
13652		1135	4.78		2.18	8.60	8.600 1.65-m antenna	Hughes 376	ANIK C3 (TELESAT-5)
13782			3.20		2.18	6.976	5 0.31-m antenna	Mitaubishi	SAKURA 2A (CS-2A)
14050			3.50		2.16	7.560		Hughes 371	GOES 6
14134	630.0		5.00		2.16	9.600	1.83-m antenna	Hughes 376	PALAPA B1
14158		1000	4.78		2.16	9.60	8.600 2.05-m antenna	Highes 376	GALAXY 1

TABLE 13
IRAS Detections of Spin-Stabilized Cylinders

								f_{12u}		$\hat{f}_{12\mu}$					
Sid	Name	$f_{12\mu}$	$f_{25\mu}$	f 60µ	range	A ^a	Tero	$f_{25\mu}$	T12/28	$f_{25\mu}$	T* 12/25	$\epsilon_{12\mu}$	E25µ	E 60µ	Ø
2608	ATS 1	0.5850	0.3550	0.000	36000	2.06	295.0	1.6479	489.50	1.9133	409.19		0.157	0.000	1.197
3029	ATS 3	0.6470	0.6480	0.2900	36000	2.60	295.0	0.9985	343.03	1.2207	342.50	0.474	0.410	0.650	0.862
3431	9 537	0.3750	0.2800	0.0000	36000	1.72	295.0	1.3393	416.02	1.6059	416.52	0.203	0.202	0.000	0.809
4353	MATO 1	1.0600	0.5450	0.0000	36000	1.11	295.0	1.9450	573.53	2.1987	571.93	0.342	0.339	0.000	4.828
4478	INTELSAT3 FO	2.5180	2.5500	1.0550	36000	1.48	295.0	0.9875	341.57	1.2161	340.23	3.334	3.307	4.200	5.898
4001	INTELSATE F2	2.1080	2.7730	1.1980	36000	6.74	295.0	0.7602	295.13	0.9553	295.30	1.108	1.099	1.292	1.112
4902	KATO 2	0.6450	0.7900	0.3400	36000	1.11	295.0	0.8165	306.46	1.0161	305.82	1.764	1.750	2.113	2.038
6052	INTELSAT4 PS	1.9670	2.6630	1.0970	36000	6.74	295.0	0.7386	290.91	0.9325	291.52	1.095	1.087	1.206	1.045
6278	ANIK AL	3.4650	4.8500	1.5800	36000	2.78	295.0	0.7144	286.09	0.9037	286.76	5.037	4.997	4.325	4.498
6437	ANIK A2	2.2250	3.3100	1.5100	36000	2.78	295.0	0.6722	277.69	0.6519	278.27	3.716	3.686	4.343	2.942
6974	DSCS 4	1.0520	1.3090	0.5870	36000	8.00	295.0	0.7574	294.58	0.9523	294.81	0.751	0.745	958.0	0.749
7250	WESTAR 1	0.7590	0.7470	0.3640	36000	2.04	295.0	1.0161	347.45	1.2488	346.14	0.680	0.675	1.023	1.289
7790	ANIK A3	0.5200	1.0600	0.6000	36000	2.78	295.0	9064.0	240.95	0.6222	241.00	1.775	1.761	2.186	0.791
1366	00ES 1	0.7560	0.7040	0.5350	36000	2.56	295.0	1.0739	359.31	1.3145	358.25	0.473	0.469	1.139	1.029
0620	INTELSAT4AF2	1.7350	2.2400	0.6050	36000	6.74	295.0	0.7746	297.95	9.9704	297.81	0.878	0.871	0.645	0.912
1638	CONSTAR 1	1.8800	1.8850	1.2050	36000	6.90	295.0	0.9973	343.60	1.2274	342.27	0.521	0.516	1.016	0.943
1112	HURISAT 2	0.8430	1.0020	0.6340	36000	3.45	295.0	0.7791	298.85	0.9751	298.60	0.823	0.817	1.314	0.864
6006	PALAPA 1	0.9600	0.7500	0.0000	36000	3.61	295.0	1.2800	401.90	1.5440	403.59	0.277	0.274	0.000	0.969
9478	MARISAT 3	0.8270	1.1460	0.6600	36000	3.45	295.0	0.7216	287.53	0.9125	288.21	0.947	0.940	1.444	0.863
9852	K1KU 2	0.3300	1.2100	0.0000	36000	1.15	295.0	0.2727	192.37	0.3367	192.38	10.004	9.924	000.0	1.809
0061	GOES 2	0.7420	0.9220	0.3600	36000	2.56	295.0	0.8048	304.06	1.0045	303.52	606.0	0.903	0.981	1.019
0143	HIMAMARI	0.7380	0.8950	0.4690	36000	3.35	295.0	0.8246	308.13	1.0277	307.43	0.654	0.649	0.958	0.771
0557	INTELSAT4AF3	1.7160	2.2110	1.0560	36000	6.74	295.0	0.7761	298.26	0.9720	298.08	0.865	0.858	1.124	0.901
8770	INTELSAT4AF6	1.0000	2.5370	1.0960	36000	6.74	295.0	0.7410	291.38	0.9351	291.94	1.040	1.032	1.202	0.998
1144	DSCS 11	1.1460	1.2050	0.5890	36000	8.00	295.0	0.8918	321.94	1.1059	320.82	0.568	0.564	954.0	0.795
1145	DSCS 12	1.6350	1.9940	0.9590	36000	8.00	295.0	0.8200	307.18	1.0222	306.52	0.903	0.975	1.319	1.146
1715	AYAMB 2	0.1800	0.7600	0.0000	36000	1.33	295.0	0.2368	183.29	0.2891	103.39	6.398	6.347	000.0	0.955
1964	00ES 4	0.7140	0.9000	0.5480	36000	3.35	295.0	0.7933	301.71	0.9910	301.27	0.691	0.685	1.155	0.751
2065	SBS 1	1.8610	2.4200	1.1990	36000		295.0	0.7690	296.86	0.9645	296.84	0.728	0.723	0.979	0.747
2295	KIKU 3	0.6250	0.6730	0.4500	36000	3.75	295.0	0.9267	329.50	1.1485	328.24	0.376	0.373	0.744	0.577
2309	COMSTAR 4	1.9670	2.6850	1.1590	36000	6.90	295.0	0.7326	289.72	0.9259	290.42	1.009	1.080	1.252	1.023
7677	HIMMARI 2	0.6160	0.7400	0.4150	36000	3.35	295.0	0.8324	309.74	1.0360	308.99	0.534	0.530	0.841	0.643
3069	WESTAR 4	1.5920	2.3040	1.0130	36000	7.80	295.0	0.6910	281.39	0.8749	282.04	0.891	0.884	1.017	0.744
3269	WESTAR S	1.7770	2.4820	1.0100	36000	7.00	295.0	0.7160	286.39	0.9055	287.06	0.917	0.910	0.992	0.822
3431	ANIK DI	1.8310	2.7370	1.1350	36000	. 84	295.0	0.6690	277.06	0.8480	277.63	0.973	0.965	1.031	0.763
3651	583 3	2.0250	2.6750	1.2450	36000		295.0	0.7570	294.51	0.9519	294.74	0.818	0.012	1.027	0.815
3652	ANIK C3	1.0710	2.6660	0.9990	36000		295.0	0.7018	203.56	0.8882	284.22	0.892	99.0	0.874	0.768
3782	SAKURA 2A	0.9210	1.1920	0.0000	36000	3.57	295.0	0.7727	297.58	0.9684	297.47	0.884	0.877	0.000	0.914
4050	9 5300	0.7320	0.000	0.5650	36000	3.35	295.0	0.9116	325.99	1.1200	324.79	0.515	0.511	1.063	0.757
4134	PALAPA B1	1.5150	2.6400	0.9800	36000		295.0	0.5739	258.14	0.7307	258.59	1.134	1.125	1.000	0.670
4150	GALAXY 1	1.6130	2.3850	1.0350	36000		295.0	0.6763	278.49	0.8570	279.10	0.836	0.829	0.933	0.670

The three INTELSAT 4A satellites with nearly equal flux densities, absorptivities, and temperatures were selected as a baseline for estimating the projected area and solar power from the 12- μ flux density. The mean value for $J_{12\mu}=1.777$ establishes $A_p=6.74/1.777$ m²/ $J_{12\mu}$. Using projected solar cell area, $A_p=6.74$ m², the mean ϵ is computed to be 0.928, and the mean α is 0.937. Thus, the α/ϵ is very close to unity, and the observed temperature, 295.9 K, is very close to that of the free-space cylinder model, which is 295 K. The INTELSAT 4A series provides a better baseline than the newer HS-376 because the analysis is simplified: the INTELSAT series did not use the new K7 high-efficiency solar cells. Table 14 gives the COMSAT model temperature for the INTELSAT 4 solar panels.

TABLE 14 INTELSAT 4A Model Temperature

		Summer Solstice	Winter Solstice	Equinox	Eclipse
Solar Power	FWD	287 K	290 K	295 K	220 K
	AFT	283 K	290 K	293 K	194 K
Sun Shield	DISC	285 K	255 K	269 K	233 K
	CONE	264 K	244 K	254 K	222 K
TWT Power		309 K	312 K	308 K	283 K
Supply		310 K	314 K	309 K	276 K

Figure 14 plots the measured (uncolor corrected) $12-\mu$ flux vs. A_p for the cylinders observed (from the data in Table 13). Also shown are parameter lines for $\epsilon = 1.0$ and 0.5, which are valid for temperatures near 295 K. Note that most of the satellites fall between $\epsilon = 1.0$ and 0.75, although several satellites, smaller than 3 m², give flux densities above the $\epsilon = 1$ parameter line. Since ϵ cannot exceed unity, the implication is that additional satellite surface area is contributing observable flux. This could be the antenna or a top-of-the-cylinder optical solar radiator. Equally likely is that the sum of the small signal and the space background yields excessive flux observations. The fact that many more objects are above the unity parameter line than below suggests that the SNR alone is not the cause. The detection threshold level, 0.15 Jy, is also shown on Figure 14. One of the smallest satellites, Ayame 2, about 1.2 m², has a

12- μ flux slightly greater than the threshold. Most of the satellites smaller than a few square meters yielded only one of two detections compared to more than a dozen for the largest. Some large cylinders also have few or a single detection such as COMSTAR 2, $A_p = 6.9 \text{ m}^2$, which appears to yield inaccurate results, e.g., T = 8000 K!

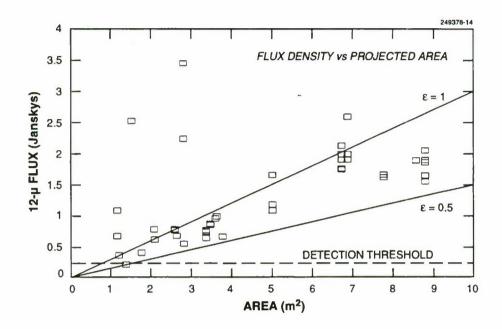


Figure 14. Flux density vs. projected area.

As might be expected, the temperature measurement suffers greatly with small flux densities and small samples. Recall the formal flux density uncertainty in the $12-\mu$ HS-376 flux data is 0.6 Jy. Figure 15 plots calculated temperature vs. the projected solar cell area for the cylinders observed. Almost all of the largest satellites are at or below 300 K. For smaller cylinders, the temperature deviation from 300 K increases. Arrows are shown for 10%, 20%, 30%, and 50% change in the J12/J25 flux ratio. Above several square meters, most satellites are close to or somewhat below 295 K.

The projected area of a geostationary spin-stabilized satellite, as well as the solar power, can be estimated directly from the flux density. The $12-\mu$ flux (uncolor corrected) is multiplied by a constant for area and a constant for power. The values of the constants for satellites of the INTELSAT 4, 4A epoch and earlier are given in Table 15. Values for power estimation are shown for beginning of life (BOL) as well as end of life (EOL). The solar power is based on 520 W available at EOL, which is seven years. Values are also given for the HS-376 class and for the somewhat different Hughes Galaxy 1.

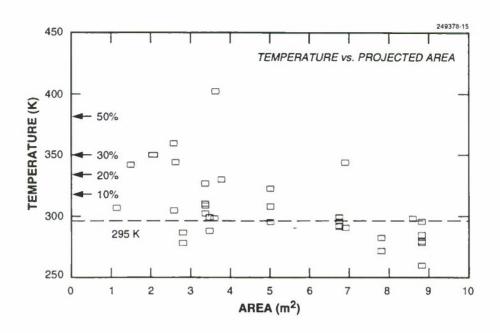


Figure 15. Temperature vs. projected area.

TABLE 15
Estimators for Geosynchronous Area and Power

	Older Cells	HS-376	Galaxy 1
*	INTELSAT 4,4A	K7 Cells	K4¾ & K7 Cells
Area/J _q ^{12µ}	3.79 m²/Jy	4.65 m ² /Jy	5.48 m²/Jy
BOL Power/J _q ^{12µ}	365 W/Jy	512 W/Jy	614 W/Jy
EOL Power/J _q ^{12µ}	292 W/Jy	410 W/Jy	491 W/Jy

Table 16 applies the simple flux density relationship to a number of satellites including Lincoln LES-6 (3431), launched in 1968. The area and power estimates for the GOES satellites 1, 2, 3, and 4 are also given together with the Japanese Himawari 1 and 2. (All are components of the global Earthwatch weather system that forecasters display on television.) GOES 1 and synchronous meteorological satellite (SMS-3) are the series built by Philco Ford for NASA before Hughes began the GOES series. The area

calculated for Table 16 agrees with the drawings: the power estimate has not been verified. Both COMSAT and MARISAT appear to yield good power and area agreement. COMSTAR may have some new solar cells that yield additional power.

Most of the smaller cylinders are older than the INTELSAT 4 series and should be near the 300-K temperature. If there is no automatic shutdown, power on the order calculated may still be available since the degradation rate is slowed after ten years (5% in the first year, 18% after ten years).

TABLE 16
Geosynchronous Area and Power Estimation

		Projected	Area (m²)	Pow	er (W)
Satellite	J _y ¹² μ	IRAS	Drawing	IRAS	Referenced
LES-6 3431	0.375	1.43	1.72	110	220 BOL
					190 1 Yr
GOES 1 8366	0.756	2.87	2.56	221	•
GOES 2 10061	0.742	2.82	2.54	216	•
GOES 4 11964	0.714	2.71	3.35	209	•
GOES 6 14050	0.732	2.78	3.35	214	*
Himawri1 10143	0.738	2.80	3.35	214	225
Himawri2 12677	0.616	2.34	2.35	180	225
COMSTAR 4 12309	1.97	7.47	6.9	575	760 BOL
					608 7Yr
Marisat 3 8882	0.882	3.35	3.45	258	220 BOL
					264 EOL
Marisat 2 9478	0.827	3.14	3.45	242	220 BOL
					264 EOL
* Not found					

Two important thresholds exist for space surveillance. The first, and lower, is the minimum signal required for detection for a metric position measurement. The second, requiring greater signal strength, is the threshold for useful, repeatable space object information such as emissivity area product and temperature. The data in Table 13 shows that cylinders smaller than about 3 m² usually exhibit flux densities below 0.6 Jy, the formal uncertainty determined in the HS-376 analysis. Low flux densities correspond to low SNRs, hence derived temperatures far from 300 K and emissivities near 0 or greater than 1, and unlikely values for α .

The demonstrated IRAS geosynchronous satellite threshold is slightly below 1 m². The threshold for repeatable space object information appears to be slightly above the 3-m² cylinder projected area corresponding to about 0.5 Jy. However, the information threshold is strongly dependent on the number of observations.

10. ROCKET BODIES

There are 668 IRAS detections of rocket bodies. The common denominator among rocket bodies is their white paint (TiO₂) covering, which helps to maintain a low temperature while on the launch pad. As discussed in Section 3, the low temperature results from the small value of $\alpha / \epsilon = 0.22$ for TiO₂. The IRAS data on rocket bodies is represented in Figure 16, which displays the $f_{12\mu}/f_{25\mu}$ temperature plotted against the $f_{25\mu}/f_{60\mu}$ temperature for all individual hits. Figure 17 displays the 107 detections on satellites for which there were more than two detections (n > 2). There are three notable features. First is the large scatter in both temperatures. The scatter of these data is significantly greater than, say, the spin-stabilized cylinders. A contributing factor could be the slow passage across the focal plane (≈ 10 sec) of a tumbling target. The change in projected area with time can introduce error in the deduced flux ratio. The extreme values are as low as 50 K and as high as 800 K. Second is the value of the median temperature (from the n > 2 data) $T(f_{12\mu}/f_{25\mu}) = 182$ K. One would expect $(\alpha/\epsilon)^{1/4} \times 295$ K = 202 K for a tumbling cylinder covered with TiO2. This is consistent with the white paint covering. Third is the difference between $T(f_{12\mu}/f_{25\mu}) = 182$ K and $T(f_{25\mu}/f_{60\mu}) = 235$ K. The median temperatures for all the detections are 283 and 249 K, respectively. This suggests that the emissivity of white paint has some wavelength dependence. This is similar to the general result (Section 7), specifically with the results from spin-stabilized cylinders (Section 8), and the discussion in Section 4—but in the opposite sense. Finally, Figure 18 gives the temperature as a function of SSC number, and Figure 19 gives the same plot for satellites with more than two detections. This is the order of launch: smaller SSC numbers are associated with earlier launches. A change in the temperature is not apparent. Dow's suggested increase in α with time would result in higher temperatures for earlier launches. This is not observed in the IRAS data.

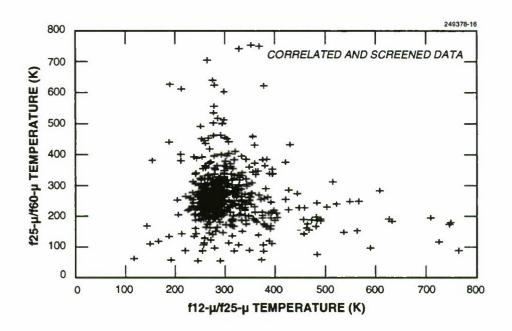


Figure 16. Rocket body temperatures.

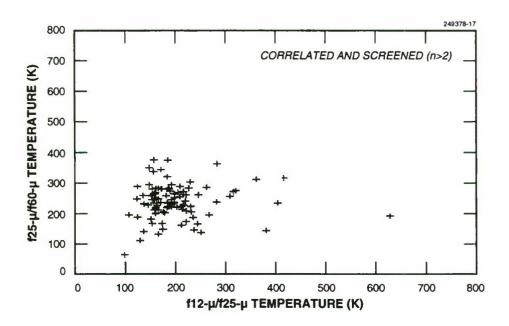


Figure 17. Rocket body temperatures, n > 2.

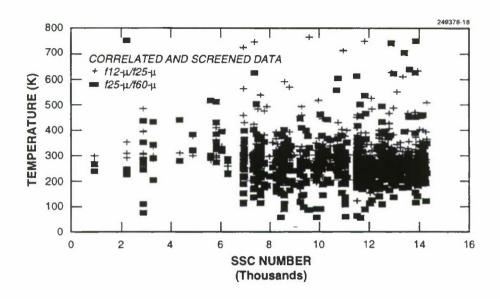


Figure 18. Rocket body temperature vs. SSC number.

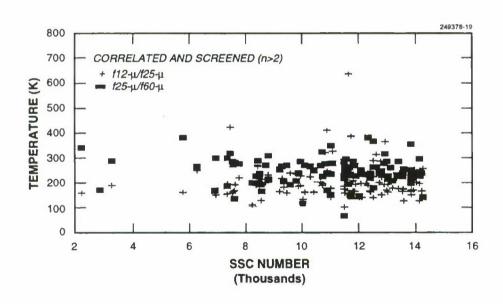


Figure 19. Rocket body temperature vs. SSC number, n > 2.

11. THREE-AXIS STABILIZED SATELLITES

11.1 GPS

The Global Positioning System (GPS) provides a means of determining position. These three-axis stabilized satellites are in 12-h circular orbits. The Block I series, Figure 20 [28], have solar panels furnishing 400 W of power and are in orbits with an inclination of 63.5°. The Block II series, Figure 21, have solar panels furnishing 700 W of power and are in orbits with an inclination of 56°.

The Block I solar panels are in the shape of a cylindrical section, whereas the Block II solar panels are flat. The satellite attitude is continuously adjusted so the antenna-populated side of the satellite is facing the earth, and the solar panel arrays are normal to the sun direction. For the Block I satellites, the earth-facing side has an area of $A_{\text{flat}} = 1.35 \text{ m}^2$, and the solar panels have a projected area of $A_{\text{solar}} = 5.28 \text{ m}^2$. To complete the model, assume the main body of a GPS satellite is cylindrical in shape. Therefore, the side silhouette is a rectangle of area $A_{\text{side}} = 1.6 \text{ m}^2$. Further, assume that this cylinder and flat solar panel model is a simple model that has no shadowing.

The IRAS satellite made 22 detections of five Block I satellites: SSC numbers 10684, 11054, 11141, 11690, and 11783. These measurements were made with sun-GPS-IRAS phase angles ranging from 78° to 120°. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 22 as a function of time. There are some significant outliers. The median temperature is 297 K and is preferred to the average to avoid a bias from the extreme values. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 23 as a function of phase angle. Except for three outliers, the temperature seems to be independent of phase angle. This suggests that the solar panels have the same temperature as the body. The reported 12- μ flux as a function of phase angle is plotted in Figure 24. A dependence of flux on phase angle is shown. Averaging groups of observations, one observes a minimum of 1.07 Jy at $\theta = 90.6^{\circ}$, 1.57 Jy at $\theta = 101.8^{\circ}$, raising to 2.71 Jy at $\theta = 120.8^{\circ}$ and 1.72 Jy at $\theta = 81.3^{\circ}$.

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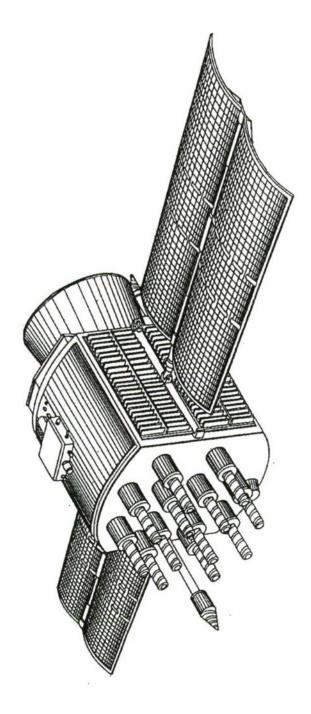


Figure 20. GPS Block I satellite.

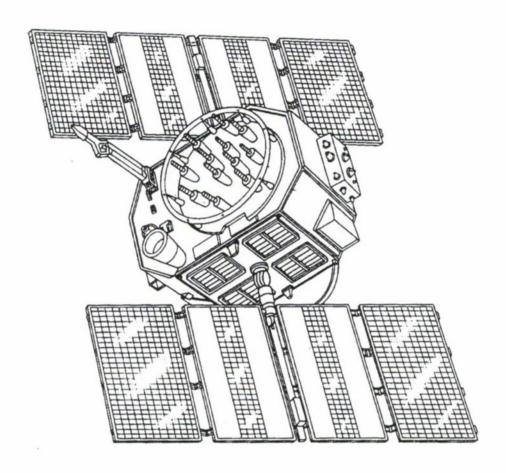


Figure 21. GPS Block II satellite.

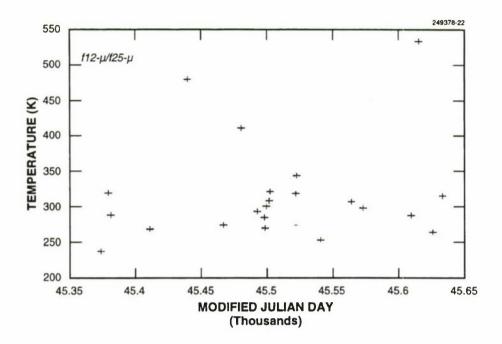


Figure 22. GPS temperature vs. date.

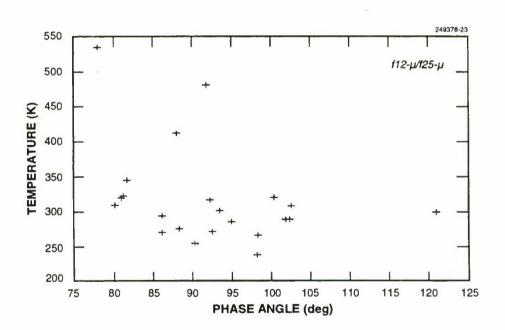


Figure 23. GPS temperature vs. phase angle.

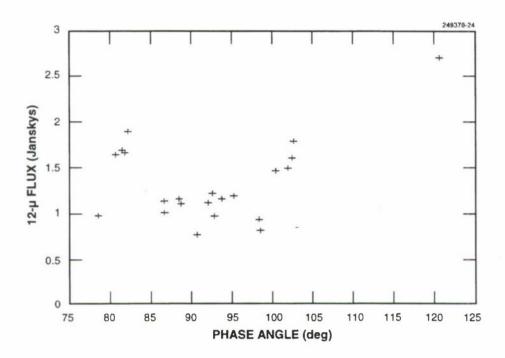


Figure 24. GPS 12-µ flux vs. phase angle.

As an illustration, the emissivity is calculated assuming the $\theta = 90^{\circ}$ data all from the earth-facing, 1.35 m² side of the GPS satellite is the only source of radiant energy. So the radiant flux density at 300 K is

$$\mathcal{F}_{12\mu}$$
(298) = 27.39 W/ μ /(effective eA) .

The IRAS color correction adjusts the 90° value to

$$J_{obs}^{12\mu} = J_{g}^{12\mu}/K_{12\mu} = 1.07/0.92 = 1.163 Jy$$
,

which is equal to $F_{\text{obs}}^{12\mu} = 2.421 \times 10^{-14} \text{ W/(m}^2 \mu)$. Now at a typical range of 19,000 km, the observed spectral radiation intensity is $\pi r^2 F_{\text{obs}}^{12\mu} = \pi (19.0 \times 10^6)^2 (2.421 \times 10^{-14}) = 27.46 \text{ W/}\mu$. Therefore, the effective area ϵA is

$$eA = \frac{\pi r^2 F_{obs}^{12\mu}}{\mathscr{F}_{12\mu}(298)} = 1.0025 \quad (m^2)$$

Since the projected area is $A = 1.35 \text{ m}^2$, the emissivity of the flat, antenna-populated side is $\epsilon = 0.74$.

A more detailed model can be defined as follows. At phase angles $\theta \neq 90^{\circ}$ and aspect angles $\phi \neq 0$, the solar panel and the satellite body, assumed cylindrical, will contribute to the observed flux. Now the cylinder and plate model will have the projection of the solar panel on the IRAS line of sight as $A_{\text{solar}}|\cos(\theta)|$. So if the cylinder flat end area is $A_{\text{flat}} = 1.35 \text{ m}^2$, the cylinder side projected area is $A_{\text{side}} = 1.6 \text{ m}^2$, and the solar panel area is $A_{\text{solar}} = 5.28 \text{ m}^2$, then the flux received would be

$$F_{obs}^{\lambda_o} = \frac{1}{\pi R^2} \left[\epsilon_{body} (A_{flat} \cos{(\phi)} + A_{side} | \sin{(\phi)} |) \mathcal{F}_{\lambda_o} (T_{body}) \epsilon_{solar}^{\pm} A_{solar} | \cos{(\theta)} | \mathcal{F}_{\lambda_o} (T_{solar}) \right],$$

where ϕ is the IRAS-GPS aspect angle. The back of the solar panel is observed at $\theta > 90^{\circ}$ and the emissivity is ϵ^{-} ; the front of the solar panel is observed at $\theta < 90^{\circ}$ and the emissivity is ϵ^{+} . The illustrative calculation can be generalized to obtain the emissivity of the solar panels accounting for the phase angle, the aspect angle, and the range. The linear regression results are summarized in Table 17.

TABLE 17
Emissivity for GPS Satellites

θ	Element	A (m²)	Э
=90	Body Flat	1.35	0.622±0.04
<90	Solar Cell Front	5.28	0.933±0.10
>90	Solar Cell Back	5.28	0.619±0.08

The emissivity for the solar panel front side, $\epsilon = 0.93 \pm 0.1$, is larger than expected. It depends directly on the adopted area. Based on laboratory measurements one expects emissivity values about 0.80. The spin-stabilized cylinders give values of solar cell emissivity between 0.6 and 0.7. The solar panel back and the main body have the same emissivity, $\epsilon = 0.62$. The two values are equal within the statistical uncertainty.

11.2 GORIZONT

The Gorizont satellites, for telephone and international television, were launched by the Soviet Union. Gorizont I (SSN 11158) has a 24-h period and was launched in 1978 into an orbit with 11.3°

inclination. By 1993 the inclination increased to more than 21°. The other Gorizont satellites are geosynchronous satellites in low inclination orbits.

The Gorizont is a three-axis stabilized satellite (see Figure 25) [28]. The main body is about 5 m long and 2 m in diameter. There are a number of appendages that will increase the effective area. Two panels of solar cells of unknown size provide power.



Figure 25. Gorizont satellite. (Used with permission. Donald H. Martin, Communication Satellites 1958–1992, ©1991, The Aerospace Corporation.)

The IRAS satellite made 26 detections of four Gorizont satellites: SSC numbers 11158, 11440, 13092, and 13624. These measurements were made with sun-Gorizont-IRAS phase angles ranging from 60° to 117°. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 26 as a function of time. The median temperature is 280 K and is preferred to the average to avoid a bias from the extreme values. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 27 as a function of phase angle. The trend is temperature independent of phase angle. This suggests that the solar panels have the same temperature as the body. The reported 12- μ flux as a function of phase angle is plotted in Figure 28. Here, a minimum of 1.3 Jy is observed at $\theta = 90^{\circ}$, raising to 2.2 Jy at $\theta = 80^{\circ}$, and 2.8 Jy at $\theta = 115^{\circ}$. Note also that none of the data for Gorizont I follow this trend. Therefore, it is not included in the following calculation.

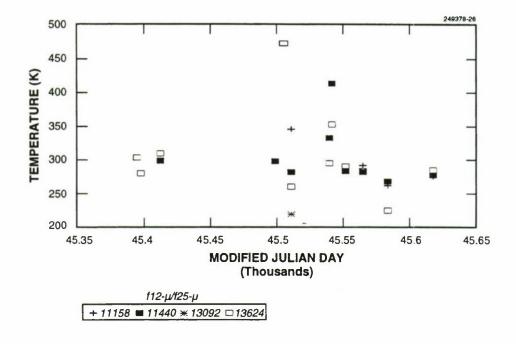


Figure 26. Gorizont temperature vs. date.

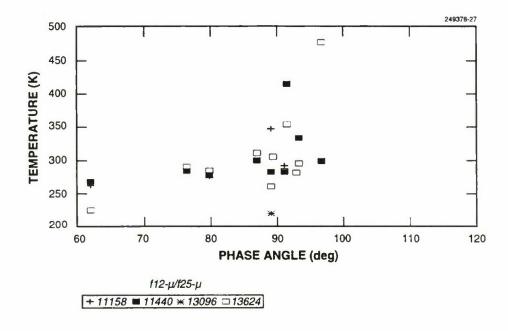


Figure 27. Gorizont temperature vs. phase angle.

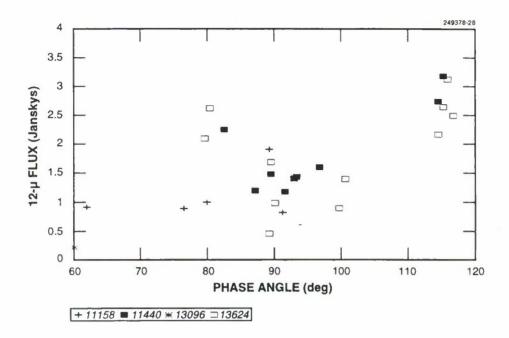


Figure 28. Gorizont 12-µ flux vs. phase angle.

Now the radiant flux density at 280 K is

$$\mathcal{F}_{12\mu}$$
(280) = 21.06 W/ μ /(effective ϵA).

The IRAS color correction adjusts the 90° value to

$$J_{obs}^{12\mu} = J_q^{12\mu}/K_{12\mu} = 1.3/0.90 = 1.44 \ Jy \ ,$$

which is equal to $F_{\rm obs}^{-12\mu} = 3.007 \times 10^{-14} \ {\rm W/(m^2\mu)}$. Now at a range of 36,000 km, the observed spectral radiation intensity is $\pi r^2 F_{\rm obs}^{-12\mu} = \pi (36.0 \times 10^6)^2 (3.007 \times 10^{-14}) = 122.4 \ {\rm W/\mu}$. Now the effective area ϵA is

$$eA = \frac{\pi r^2 F_{obs}^{12\mu}}{\mathscr{F}_{12\mu}(280)} = 5.81 \quad (m^2)$$

In this case we chose to adopt a value of the emissivity, $\epsilon = 0.70$, and calculate the earth-facing area, which is $A = 8.3 \text{ m}^2$. This is equivalent to a 3.2-m-diameter main body, which is 60% larger than the nominal main body dimension. We can estimate the size of the solar panels. Following the same

reasoning used for GPS (see Section 11.1), the projection of the solar panel on the IRAS line of sight is $A_{\text{solar}}|\cos(\theta)|$. So if the cylinder flat end area is A_{flat} , the cylinder side project area is A_{side} , and the solar panel area is A_{solar} , then the flux received would be

$$F_{obs}^{\lambda_o} = \frac{1}{\pi R^2} \left[e_{body} (A_{flat} \cos(\varphi) + A_{side} | \sin(\varphi) |) \mathcal{F}_{\lambda_o} (T_{body}) e_{solar} A_{solar} | \cos(\theta) | \mathcal{F}_{\lambda_o} (T_{solar}) \right],$$

where ϕ is the IRAS-Gorizont aspect angle. The back of the solar panel is observed at $\theta = 115^{\circ}$, and the front is observed at $\theta = 80^{\circ}$. The same regression calculation can be done. Since we have no prior knowledge of the solar plane area, we chose to determine the emissivity area product ϵA . The results are summarized in Table 18.

TABLE 18
Emissivity Area for Gorizont Satellites

θ	Element	€A (m²)	A (m²)
=90	Body Flat	4.38±0.6	6.25
<90	Solar Cell Front	30.75±6.8	33
>90	Solar Cell Back	15.50±2.2	25

This estimate of the earth-facing area, $\epsilon A = 4.38$, leads to an area A = 6.25 m², equivalent to a main body diameter of 2.8 m, which is about 40% larger than the nominal. The estimates of the panel size from the front and back emissions differ due to measurement error, model error, and above all, error in the adopted emissivity. Based on the GPS results one might expect the back panel emissivity to be lower by 30% to 50%. If we adopt the GPS values of emissivity from the body and solar cells, then we get values given in Table 18. The solar panel area is of the order of 25 to 35 m². This is consistent with two solar panels, each providing 1 to 2 kW of power.

11.3 RADUGA

The Raduga satellites were the first Soviet satellites in geosynchronous orbit. They were three-axis stabilized and were used for telephone, telegraph, and television transmission. The characteristics are

similar to the Gorizont satellites (Section 11.2). The IRAS satellite made 27 detections of six Raduga satellites: SSC numbers 10159, 10987, 11708, 12897, 13669, and 13974. These measurements were made with the sun-Raduga-IRAS phase angle ranging from 84° to 103°. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 29 as a function of time. These measurements were made with sun-Raduga-IRAS phase angles ranging from 76° to 103°. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 30 as a function of phase angle. The median temperature is 288 K and is preferred to the average to avoid a bias from the extreme values. The trend is temperature independent of phase angle. This suggests that the solar panels have the same temperature as the body. The reported 12- μ flux as a function of phase angle is plotted in Figure 31. The same regression calculation that was done for GPS and Gorizont satellite data can be done. Since there is no prior knowledge of the solar plane area, we determined the emissivity area product ϵA . The results are summarized in Table 19.

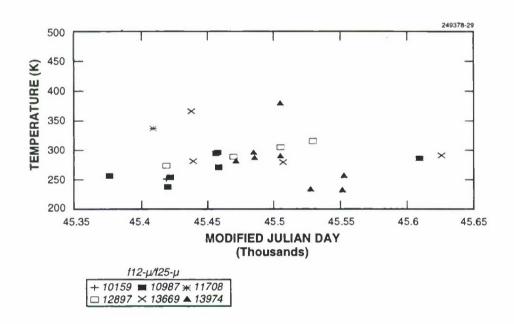


Figure 29. Raduga temperature vs. date.

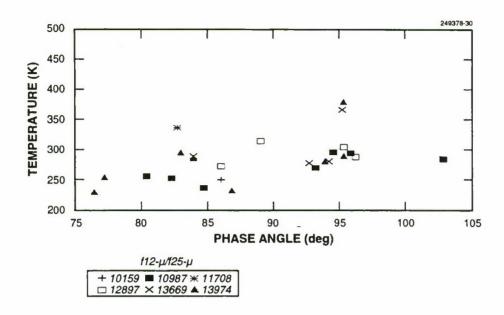


Figure 30. Raduga temperature vs. phase angle.

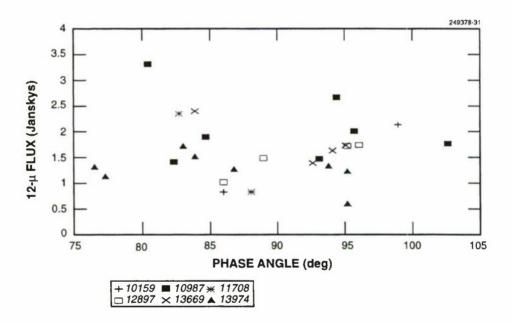


Figure 31. Raduga 12-µ flux vs. phase angle.

TABLE 19
Emissivity Area for Raduga Satellites

θ	Element	€A (m²)	A (m ²)
=90	Body Flat	2.80±1.4	4.00
<90	Solar Cell Front	37.13±14.4	40
>90	Solar Cell Back	36.44±16.4	58

This estimate of the earth-facing area, $\epsilon A = 2.8 \text{ m}^2$, leads to an area $A = 4 \text{ m}^2$, which is equivalent to a main body diameter of 2.25 m. There is no prior knowledge of the body size. The estimates of the panel size from the front and back emissions differ due to measurement error, model error, and above all, error in the adopted emissivity. Based on the GPS results one might expect the back panel emissivity to be lower by 30% to 50%. However, the ϵA product for the front and back are nearly equal. Nevertheless, uncertainty is quite large. If the GPS values of emissivity from the body and solar cells are adopted, then values for A are as given in Table 18. The solar panel area is larger than 35 m². This is consistent with two solar panels, each providing 1 to 2 kW of power.

11.4 EKRAN

The Ekran satellites were three-axis stabilized and were used for telephone, telegraph, and television transmission within the USSR and bordering countries. The characteristics are similar to the Gorizont satellites (Section 11.2), Figure 32 [28], with the addition of a complex antenna array on the earth-facing side of the satellite. The IRAS satellite made 27 detections of five Ekran satellites: SSC numbers 11273, 12120, 12564, 13554, and 13878. These measurements were made with the sun-Ekran-IRAS phase angle ranging from 70° to 97°. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 33 as a function of time. Temperatures determined with the $f_{12\mu}/f_{25\mu}$ flux ratio are plotted in Figure 34 as a function of phase angle. The median temperature is 284 K and is preferred to the average to avoid a bias from the extreme values. There is a suggestion of an increase in temperature for phase angles greater than 90°. The reported 12- μ flux as a function of phase angle is plotted in Figure 35. The temperature measurements are consistent with those of other satellites. However, there seems to be no simple relation between phase angle and flux. A more complex radiation model is necessary to explain these data, which are probably due to the large antenna array.

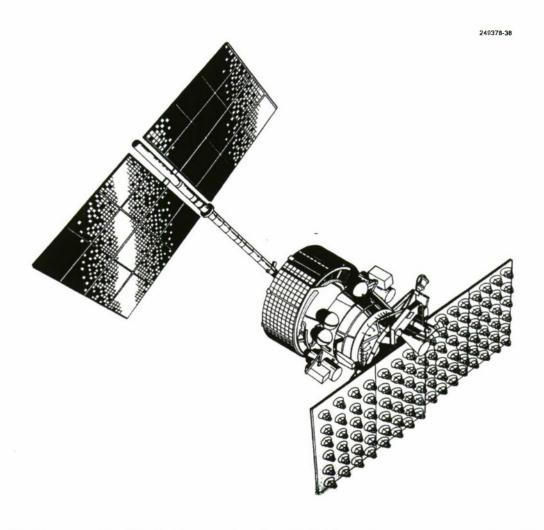


Figure 32. Ekran satellite. (Used with permission. Donald H. Martin, Communication Satellites 1958–1992, ©1991, The Aerospace Corporation.)

11.5 ATS-6

The Application Technology Satellite (ATS)-6, SSC number 7318, was active for more than five years. It was used for a number of experiments in satellite communications technology. It was a three-axis stabilized satellite with an enormous parabolic antenna, a split cylindrical solar cell panel, and a rectangular payload; see Figure 36 [28].

The IRAS satellite made three detections of the ATS-6, which are given in Table 20. The curiosity is the large flux measured. The antenna material was copper netting on a structure of 48 aluminum ribs, of which there is no specific knowledge of the emissivity or temperature. Still, even with extreme values of the emissivity, it is difficult to calculate such a large flux.

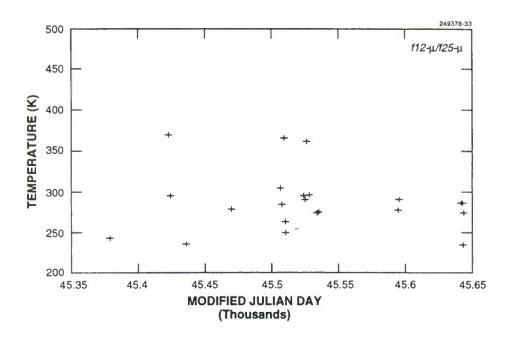


Figure 33. Ekran temperature vs. date.

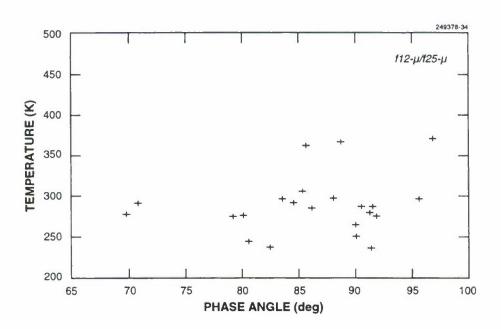


Figure 34. Ekran temperature vs. phase angle.

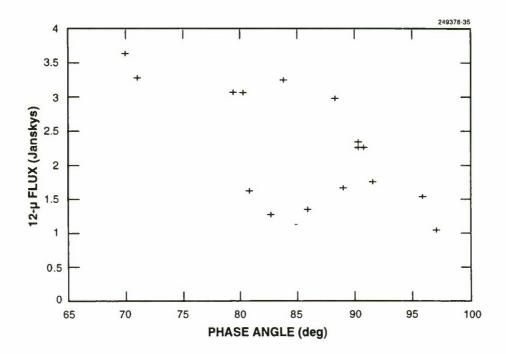


Figure 35. Ekran 12- μ flux vs. phase angle.



Figure 36. ATS-6 satellite. (Used with permission. Donald H. Martin, Communication Satellites 1958–1992, ©1991, The Aerospace Corporation.)

TABLE 20
ATS-6 Observed Flux

MJD	J _a ^{12µ}	J _a ^{25µ}	J ₂ 60µ	Т _{12ш/25µ}	T _{251/60µ}	θ	ф
45521.605	4.88	6.98	2.24	284	349	92	0.3
45551.092	5.48	7.45	2.24	291	379	96	1.8
45641.552	5.51	9.00	3.42	267	287	89	0.8

12. DISCUSSION

The IRAS satellite made both position and radiometry observations of RSOs. The accuracy of the position measurement was limited by the detector size of 4.5 arcmin. This was adequate for correlation with a satellite catalogue to associate observations with specific satellites. Use of these position observations for determination of ephemerides is not addressed here. Concerning the radiometry measurements, the accuracy of a single observation is experimentally determined to be ± 0.6 Jy. Of the 190,000 detections, 2047 were correlated with the known satellite catalogue. The correlation used both the position and velocity measurement. The observations were correlated with 452 satellites. Due to the zenith pointing of IRAS, most of the observations are on deep space satellites, many of which are geosynchronous.

The radiometer flux data required some screening. The principal issue concerned partial detections where the image crossed the detector edge. Lacking detailed knowledge of the image path across the detector, this could only be done statistically. Therefore, even the screened data will have some partial detections. Nevertheless, screening individual detections significantly improved the data consistency.

Calibration of the IRAS data remains a nagging question. There have been suggestions about revision of the IRAS calibration, as reviewed in Section 5. There are significant questions of interpretation that would be affected by a change in calibration. For example, for all the IRAS data there is the uniformly lower temperature observed with the 25- to $60-\mu$ band flux ratio than observed with the 12- to $25-\mu$ band flux ratio. This is true for the majority of the observations and is particularly striking in the Hughes HS-376 cylinders. As Table 3 shows, most critiques suggest that the $25-\mu$ fluxes are high by 4% to 10%. For our analysis of the HS-376 cylinders, changing the calibration in this way would result in worse agreement between the IRAS observations and the models. Also, any decrease in the $25-\mu$ flux would increase the disagreement between the $12-\mu/25-\mu$ and the $25-\mu/60-\mu$ temperatures and would increase the derived emissivities. On the other hand, for both classes, the HS-376 and the INTELSAT 4 and 4A satellites, the average value measured for emissivity is just below unity. This is 5% to 15% above the laboratory measurements for solar cells. Perhaps this is due in part to the simple model ignoring the antenna heat. Perhaps some of the excess flux density should be attributed to calibration error on the order of 5% too high in both the 12- and 25-\mu bands. How to apportion these disagreements between dependence of solar cell emissivity on wavelength and calibration remains undecided. Therefore, no definite conclusion about IRAS calibration can be reached from this analysis.

IRAS observations were serendipitous. Therefore, there is no systematic set of observations for analysis. On average there were six observations for each satellite, but many satellites were observed only once. This was ameliorated by using classes of satellites that can be assumed to be virtually identical, for example, the Hughes HS-376 spin-stabilized cylinders, the GPS satellites, and a number of Soviet communication satellites. In this case a data set exceeding 20 detections could be formed.

The discussion of IR satellite radiometry inevitably turns on the properties of spacecraft materials. These include paints, solar cells, antennas, and radiators. Existing laboratory data is suggestive but not conclusive. Also, data on the effects of the space radiation environment on spacecraft materials from the LDEF are helpful. Nevertheless, the IRAS data must stand on its own. For example, the effects of TiO₂ on

rocket body temperature is consistent with expectations. However, expectations that ${\rm TiO}_2$ will change absorptivity after exposure to the space environment are not supported by the IRAS data.

With these caveats, one can see a lot of information in the IRAS radiometry. Even with very simple models one can learn about the absorptivity and emissivity of spacecraft materials. Alternatively, by using the geometrical variation of phase angle and aspect angle, one can estimate the physical size of certain spacecraft components. Further progress will entail considerably more complex models and detailed knowledge of the spacecraft.

The fact that the IRAS was a space-based platform was extremely valuable. The whole question of atmospheric absorption is avoided, and the data have remarkable consistency. In addition, the multispectral data immediately provides information on the object temperature and leads directly to physical analysis. Much analysis of the ground-based data has suffered from both these limitations.

In 1992 no data on geosynchronous objects were published from either ground- or space-based LWIR sensors. In the winter of 1992–1993, Seniw and Rieke collected LWIR data. This included 60 tracks on 20 geosynchronous satellites in the N- and Q-bands, which allowed temperature measurements on two objects [3]. The data were obtained over a wide range of sun-sensor-satellite phase angles and included observations of temperature in and out of the earth shadow. Six geosynchronous objects observed by Seniw and Rieke were observed by IRAS a decade earlier. Two cylinders, GOES 6 and the Hughes Galaxy 1, had illumination phase angles of 52° and 64°, respectively. This is close enough to the IRAS near 90° phase angle to warrant comparison. Only astronomical N-band data were obtained from the ground so that temperature could not be determined. The radiant intensity, watts/steradian, was computed from the IRAS temperature and satellite ϵA_p and multiplied by the fraction of the total in N-band. Table 21 compares the University of Arizona and the IRAS measurements. In this calculation, following the logic in Section 9, the EOL temperature estimate for Galaxy 1 of 295 K is used instead of the observed value of 278 K.

Recall that the IRAS pointing logic generally viewed sun-tracking solar panels nearly edge-on. Therefore, for those satellites, the radiant intensity was probably from the equipment body and antennas, which is a small fraction of the RSO's radiant intensity at maximum phase. Slight changes in aspect angle would yield moderate changes in observed intensity, which we have exploited. On the other hand, spin-stabilized cylindrical solar panels present a near-constant geometry. The despun antennas are pointing at the earth but are edge-on to the sun.

TABLE 21
Ground-Based and IRAS Observations

	Category	Univ Arizona	IRAS
GOES-6	Phase Angle	52°	90∘
Weather Sat 14050	Reported Flux	0.74 Jy @ 10.6 μ	0.732 Jy @ 12 μ
	Temperature	-	325 K
	Radiant Inten.	- 169 W/sr	132 W/sr
Galaxy 1	Phase Angle	64∘	90∘
HS-376 14158	Reported Flux	1.58 Jy @ 10.6 µ	1.61 Jy @ 12 µ
	Temperature		295 K
4	Radiant Inten.	372	382

13. CONUNDRUMS

The IRAS debris data base presents a number of unanswered questions—some going to the heart of this analysis. In Section 9, analysis of Hughes HS-376 spin-stabilized cylinders (the simplest, homogeneous, and unambiguous data set available) is presented. The radiometric model is quite simple and reasonably good. The data set, after screening, seems consistent, and the derived results are quite reasonable. From the analysis of variance, we derived the formal uncertainty of a single observation as ± 0.6 Jy in the 12- μ band. However, to reach this state, some data had to be edited. There were far more screened detections, with errors exceeding 1.2 Jy, than expected on the basis of Gaussian statistics. One could postulate that another satellite was observed at that time and direction. If true, then where is the measurement of the correlated satellite that we know was there at that time? One must assume that the detection was indeed of the correlated satellite, with quite a large flux error. Consequently, we have considerable concern about the error statistics of the IRAS debris data base. Unfortunately, there is insufficient data to actually discuss the error distribution function.

A data set with a related problem is derived as follows. Select all the detections that were obtained with small velocity (< 6 arcmin/sec) across the focal plane. There are 22,000 such detections. Of these, 2047 were correlated with the catalogue. Excluding the pathological case of a co-orbiting satellite, these would be observations of satellites at extreme range: geosynchronous satellites or high eccentricity satellites observed at apogee. In both cases the observed range would be 36,000 km or greater. Among these data, there are 54 detections with a $12-\mu$ flux > 100 Jy and 10 exceeding 900 Jy. Note that a flux greater than 999.99 Jy is listed as 999.99. Of these, only nine had a 12- to $25-\mu$ flux ratio that implied a plausible temperature, e.g., between 250 K and 400 K. Most of the large fluxes occur in only one band. There are cases where both detectors for a band registered these high values. Only three were observed at small declination, i.e., in the geosynchronous belt, with most having very high declination. These very large flux detections seem to have no consistent orbits. One must conclude that these are all bogus observations.

The reliability of a single detection, even with multiple hits, is seriously in question. With the spin-stabilized cylinders, there were hits with twice the "correct" flux value. These can be deduced because in this case there are multiple detections of a well-understood target. Detections at a geosynchronous range of 100 Jy or more seem implausible. It is difficult to assess the observed flux without multiple detections. Therefore, we believe that any conclusions based on a single detection must be made very cautiously.

APPENDIX A IRAS DETECTIONS CORRELATED WITH THE KNOWN SATELLITE CATALOGUE

This appendix is a list of the 2047 IRAS detections correlated with the known satellite catalogue. This selection includes the detections in agreement within 0.6°. In some cases detections were not used in further analysis. The reasons include angular velocity error exceeding some value, the radiometric data being noisy, or exceeding a statistical test, as discussed in the text.

The list contains 12 items for each detection. First is the satellite identification number. We have chosen the identification given by the US Air Force Space Surveillance Center (SSC). In Appendix B we provide other information such as the international designator and orbital characteristics. All 12 items, in the following order, are given below.

- 1. Satellite identification number.
- 2. Modified Julian date of the detection.
- 3. Temperature, Kelvin, derived from the $f_{12} / f_{25} \mu$ flux ratio.
- 4. Temperature, Kelvin, derived from the $f_{25} / f_{60\mu}$ flux ratio. If the flux ratio is not available, then the temperature is reported as -1.
- 5. Quoted $12-\mu$ flux, Janskys. Data screening is applied as described, eliminating partial detections, and the selected detections are averaged to obtain the value given here.
- 6. Quoted 25- μ flux, Janskys, screened in the same way as the 12- μ data.
- 7. Quoted 60- μ flux, Janskys, screened in the same way as the 12- μ data.
- 8. IRAS-satellite-sun phase angle, degrees.
- 9. IRAS satellite aspect angle, degrees.
- 10. Satellite declination of the detection.
- 11. IRAS-to-satellite range derived from the ephemeris, km.
- 12. Observed angular rate of the detection, arcmin/sec.

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2.6500	040	C/T.	470	645	.805	.360	.650	.760	.855	.435	.060	.195	495	.875	.200	1.150	.100	2.540	2.885	.595	.835	.410	.000	.035	.960	.870	.820	.590	.740	.985	.480	.945	.550	.715	.725	.630	.920	. 595	.835	. 225	.840	405	.800
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000	640	040	650	.825	.440	475	.820	.320	.225	.350	.400	.635	.250	1.1550	.270	.975	.975	.520	.175	.135	.785	.235	.765	540	.830	.915	. 945	.675	.420	.845	.225	.475	.690	.970	.770	.925	990	970	.150	.430	595	340	.065	.520
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0.4650	.775	.485	.525	.510	.755	.320	.325	315	250	815	575	945	395	.915	.590	5.330	.170	.960	.100	.370	.945	.460	.835	.620	.350	.460	.350	.925	.275	.575	.105	.890	.475	.675	.675	.285	.420	.430	.850	810	25.5	375	450	980	
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530	000	535	.000	.000	.000	.050	.455	.765	.080	.900	.990	.220	.630	.510	.840	.640	.575	190	.160	.210	.085	.460	000	.000	000	0.3600	.605	.460	.000	.000	.610	.780	.000	000.	. 795	.105	.835	.945	.815	0.730	.715	. 925	.205	.910
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725	570	665	860	.010	675	.715	.840	.385	.065	.655	.810	.050	.675	.490	.390	.720	.865	.750	.115	.050	105	905	310	500	540	430	.535	.920	.565	.920	.610	.670	.130	.565	.055	.045	.965	.330	.015	0.940	680	535	395	1.3050
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675	0.760	460	1.2050	.360	.570	.695	.165	.595	.550	300	.935	.285	.335	.875	.720	.270	.370	.920	.310	.670	.660	.105	.430	.580	.225	.660	.450	.980	.535	.175	.690	3.785	.785	.895	.835	.255	.955	.590	.240	.225	.200	.390	.425	.200
715	0.585	800	19	110	645	.330	.270	.050	.530	.395	.570	625	.945	.940	695	.880	.165	.450	.615	305	840	950	.500	190	.810	.825	.420	300	.695	.065	585	3.500	.300	.560	.240	895	.935	745	895	.225	960	805	950	LO.
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.915	. 930	.580	.470	.010	.630	.360	.285	.770	.935	.405	.530	.580	.295	0.4350	.140	.580	.025	.355	.805	.600	.320	.075	7.730	.265	.170	.700	.865	.170	.415	.825	.810	.745	.375	1.120	.710	.340	.585	.015	.070	.040	.020	.660	.100	.600
.150	780	.685	.425	.520	.920	.650	390	.405	.605	.660	.435	.410	.575	0.3950	. 995	.560	. 285	.390	.470	.140	.070	.535	.815	.820	.170	.650	.720	.245	.335	.795	.065	.440	.340	0.700	.910	.800	.790	.665	.315	. 565	.620	.110	.450	.720
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3.5250	OTS.	.430	. 290	.600	.785	.320	745	.520	.010	945	495	6.490	.640	.640	.715	.245	.100	.165	350	340	385	440	580	890	3.905	.780	.290	.785	.230	.630	.600	.740	.840	.200	090.	.670	.310	.945	.330	.570	.010	110	.035	.270
2.9700	640	.600	.670	.005	415	.275	.785	495	790	915	030	5.070	515	.945	990	.710	.340	.255	585	565	320	920	226	810	810	960	400	.015	.420	.540	.000	.605	.220	.135	.695	.360	.755	.125	.700	.910	640	710	160	.265
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.260	.055	.350	.165	.200	00.	.645	.350	.620	.525	.980	.520	.550	.655	.305	.000	.390	000	.345	1.0600	.280	.780	.935	.830	000	.600	340	
.86	.32	.32	.40	36	40	. 22	.17	.51	1.70	.63	8.15	2.49	0.95	.61	0.45	.78	.51	. 08	2.6100	.67	.19	.52	00	45	85	9	
.305	.930	.835	.275	920	800	.245	825	240	750	.640	000	.005	705	345	520	510	450	920	1.9700	575	.810	615	540	150	870	175	
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APPENDIX B SATELLITES OBSERVED BY IRAS (AT LEAST ONCE)

This appendix identifies the 465 satellites observed by IRAS. This list contains the following information.

- 1. Air Force Space Surveillance Center identification number.
- 2. Year of launch.
- 3. International designator.
- 4. Country of origin.
- 5. Type of object: PL denotes a payload; RB, a rocket body; and DB is debris.
- 6. Status of the object.
- 7. Common name of object or a comment.
- 8. Inclination of the object, degrees.
- 9. Eccentricity of the object.
- 10. Semimajor axis of the object, earth radii.
- 11. Mean motion of the object, revolutions per day.

Sid	Year	Designation	Owner	Mission	Status	Comment	Inclination	Eccentricity	Semi major axis	Mean motion
			1	1 1 1 1 1	1					
20	1959	007A	USA	PL	NA	VANGUARD 3	33.3	.1880	1.330300	11.115000
282	1962	0128	USA	RB	NA	RANGER 4//RB (no element set)	7.	.1384	6.613023	1.002251
573	1963	013A	USA	PL	NA	TELSTAR 2	42.8	.4011	1.923095	6.392022
575	1963	013B	USA	RB	NA	TELSTAR 2//SDC 573	42.8	.4007	1.921426	6.400349
748	1964	006B	SOV	PL	NA	ELEKTRON 2	63.6	.6030	6.363943	1.061707
751	1964	Q900	SOV	RB	NA	ELEKTRON 1//SDC 746	64.0	.5809	6.450280	1.040463
830	1964	038B	SOV	PL	AC	ELEKTRON 4	62.9	.8308	6.228553	1.096366
869	1964	0490	SOV	PL	NA	COSMOS 41	71.5	.6721	4.150794	2.015250
868	1964	049E	SOV	RB	NA	COSMOS 41//SDC 869	71.5	.6720	4.163013	2.006391
1360	1965	034B	USA	PL	AC	LES-2	32.2	.3971	2.378194	4.648185
1361	1965	034C	USA	PL	NA	LCS-1	32.1	.0010	1.236000	9.890000
2222	1966	0533	USA	RB	NA	OPS 9381//SDC 2207	11.9	.0162	6.341975	1.067199
2608	1966	110A	USA	PL	NI	ATS 1	13.5	.0045	965909.9	1.003723
2643	1961	001D	USA	DB	IN	INTELSAT IIF-2//SDC 2639	26.9	.6492	2.996155	3.287286
2653	1961	003F	USA	PL	NI	OPS 9326 (IDCSP-13)	11.4	.0034	6.300318	1.077799
2868	1961	5990	USA	RB	NA	OPS 9331//SDC 2862	9.5	. 0049	6.246769	1.091676
3029	1967	1111	USA	PL	IN	ATS 3// SYNCTM 1050Z	13.1	.0024	6.610919	1.002733
3292	1968	0507	USA	RB	NA	IDCSP 4-1,6 Transtage	12.1	.0168	6.386718	1.055998
3307	1968	055A	USA	PL	IN	EXPLORER 38	120.8	.0011	1.916780	6.422126
3431	1968	0810	USA	PL	AC	LES 6//SYNCTM 1330Z	11.3	6000.	6.610243	1.002895
3432	1968	081E	USA	RB	NA	LES-6//SDC 3431	11.0	0600.	6.559110	1.014639
3848	1968	055C	USA	DB	NA	EXPLORER 38//SDC 3307	120.8	.0014	1.916242	6.424930
3954	1969	046D	USM	PL	AC	VELA//OPS 6909	61.6	.2914	18.456831	.214949
3955	1969	046E	USM	PL	AC	VELA//OPS 6911	61.1	.3332	18.452632	.215107
4353	1970	021A	NAT	PL	NA	NATO I	11.3	.0002	6.611288	1.002646
4354	1970	0218	USM	RB	NA	NATO I//SDC 4353	26.1	.7008	3.595968	2.500028
4366	1970	A720	USM	PL	AC.	VELA//OPS 7033	61.2	.0849	18.441386	.215341
4368	1970	027B	USM	PL	AC.	VELA//OPS 7044	57.4	.0668	18.472443	.214696
4478	1970	055A	ITS	PL	N.	INTELSAT IIIF-8	13.1	.0336	6.524887	1.022637

Sid	Year	Designation	Owner	Mission	Status	Comment	Inclination	Eccentricity	Semi major axis	Mean motion
			1	1 1 1 1 1	1 1 1 1 1		1 1 1 1 1 1 1 1			
4630	1970	093A	USA	PL	IN	OPS 5960	15.7	.1366	5.858588	1.201954
4881	1971	006A	ITS	PL	NA	INTELSAT IVF-2	10.6	.0037	6.674583	.988434
4882	1971	006B	USA	RB	NA	INTELSAT IVF-2//SDC 4881	27.4	.7192	3.911328	2.203818
4902	1971	W600	NAT	PL	NI	NATO IIB//DO NOT ATTEMPT TO T	11.6	.0211	6.611293	1.002657
4925	1964	0868	USA	DB	NA NA	Explorer 26 Fragments	19.8	.5095	2.141244	5.442057
5204	1971	0398	NSM	PL	NA	OPS 3811 MEWS 2	8.6	.0008	6.706388	.981406
5205	1971	039B	USM	DB	NA	OPS 3811 MEWS 2//SDC 5204	9.2	.0111	6.604062	1.004298
5589	1971	0950	USA	RB	NA	DSCS 1//SDC 5587	12.0	.0165	6.750073	.971874
5598	1971	M960	USA	PL	NA	EXPLORER 45	3.2	.5734	2.444160	4.462531
5816	1972	003B	USA	RB	NA	INTELSAT IVF-4//SDC 5775	28.3	.7185	3.910264	2.204621
5851	1972	0108	MSD	PL	AC	OPS 1570 IMEWS 3	.2	.0060	6.595500	1.007063
5991	1969	2690	USA	RB	NA	ATS 5//SDC 4068	17.2	.6703	4.024681	2.111305
6052	1972	041A	ITS	PL	NA	INTELSAT IVF-5	8.5	.0003	6.618852	1.000938
6058	1972	041B	USA	R.B	NA	INTELSAT IVF-5//SDC 6052	27.2	.7229	3.901311	2.212246
6192	1972	072A	SOV	PL	IN	COSMOS 520	7.79	.6219	4.153195	2.013576
6278	1972	W060	CAN	PL	NA	ANIK A1	8.5	.0024	6.675102	.988313
6302	1972	072E	SOV	RB	NA	COSMOS 520//SDC 6192	9.79	.6170	4.120567	2.037535
6437	1973	023A	CAN	PL	NA	ANIK A2	7.2	.0007	6.631945	.997984
6691	1973	0408	USA	PL	AC	OPS 6157 MEWS 4	7.2	6000.	6.630756	.998236
6779	1967	001X	USA	DB	NA	INTELSAT IIF-2//SDC 2639	28.1	.7178	3.923480	2.193502
6877	1973	076A	SOV	PL	AC	MOLNIYA 2-7	64.2	.7540	4.183995	1.991358
6893	1973	078A	USA	PL	AC	IMP 8	47.3	.1234	35.158855	.081754
6898	1973	D920	SOV	RB	NA	MOLNIYA 2-7//SDC 6877	62.9	.7410	4.223020	1.964020
9169	1973	084A	SOV	PĽ	IN	COSMOS 606	67.9	.6489	4.166774	2.003682
66939	1973	084D	SOV	RB	NA	COSMOS 606//SDC 6916	67.6	.6502	4.119941	2.037931
6958	1973	A760	SOV	PL	AC	MOLNIYA 1-26	64.0	.7378	4.163324	2.006199
6974	1973	100B	MSD	PL	AC	OPS 9434 DSCS 4	10.9	.0063	6.611143	1.002694
9269	1973	100D	USA	RB	NA	OPS 9433-9434//SDC 6973	11.7	.0271	6.850786	.950524
7000	1973	106A	SOV	PL	AC	MOLNIYA 2-8	63.9	.7476	4.162363	2.006889
7178	1973	097D	SOV	RB	NA	MOLNIYA 1-26//SDC 6958	64.2	.7354	4.228946	1.959730

Sid	Year	Designation	Owner	Mission	Status	Comment	Inclination	Eccentricity	Semi major axis	Mean motion
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7250	1974	022A	USA	PL	NA	WESTAR 1	6.8	5000.	6.627711	. 998937
7260	1974	023A	SOV	PL	AC	MOLNIYA 1-27	64.1	.7509	4.157048	2.010475
7264	1974	023E	SOV	RB	NA	MOLNIYA 1-27//SDC 7260	64.1	.7553	4.205795	1.975901
7276	1974	026A	SOV	PL	NA	MOLNIYA 2-9	62.6	.7363	3.900813	2.212090
7318	1974	039A	USA	PL	NA	ATS 6	10.7	.0030	6.536818	1.019838
7324	1974	039C	USA	RB BB	NA	ATS 6//SDC 7318	10.9	.0021	6.593466	1.006711
7354	1974	0500	SOV	82	NA	COSMOS 665//SDC 7352	62.4	.7464	4.031656	2.105105
7369	1974	054A	USA	PL	A C	OPS 7518 NTS1 (TIMATION 3)	125.1	6200.	3.133876	3.072146
7372	1973	106D	SOV	82	NA	MOLNIYA 2-8 // SDC 7000	64.2	.7384	4.224188	1.962997
7373	1974	026E	SOV	RB	NA	MOLNIYA 2-9//SDC 7276	63.0	.7539	4.213054	1.970793
7376	1974	056A	SOV	PL	IN	MOLNIYA 2-10	61.8	.7282	4.163997	2.005808
7382	1974	056D	SOV	RB	NA	MOLNIYA 2-10//SDC 7382	62.3	.7446	4.218274	1.967182
7468	1974	075C	USA	828	NA	WESTAR 2//SDC 7466	24.4	.5651	2.368725	4.676869
7480	1974	0818	SOV	PL	DD	MOLNIYA 1-28//29 DEC 1985	63.4	.7444	3.968185	2.154764
7485	1974	081D	SOV	QQ	NA	MOLNIYA 1-28//SDC 7480//27 OC	63.1	.7501	4.030396	2.106265
7540	1974	092A	SOV	PL	DD	MOLNIYA 3-1//15 MAY 1986	62.9	.1347	1.178695	13.290740
7545	1974	0938	USA	RB	NA	INTELSAT IVF-8//SDC 7544	25.5	.7235	3.908513	2.206224
7546	1974	092E	SOV	DB	NA	MOLNIYA 3-1//SDC 7540	64.0	.7555	4.194397	1.983940
7583	1974	102A	SOV	PL	QQ	MOLNIYA 2-11//07 JUL 1988	61.9	.3008	1.453230	9.727411
7586	1974	102D	SOV	RB	NA	MOLNIYA 2-11//SDC 7583	62.0	.6616	2.997658	3.283689
7625	1975	A100	SOV	PL	NA	COSMOS 706	67.8	. 5069	4.160392	2.008410
7629	1975	007D	SOV	R3	NA	COSMOS 706//SDC 7625	9.19	.5280	4.159821	2.008818
7641	1975	009A	SOV	PL	QQ	MOLNIYA 2-12//04 JUL 1985	63.9	.7379	3.889347	2.221856
7653	1975	Q600	SOV	RB	NA	MOLNIYA 2-12//SDC 7641	63.9	.7455	3.998651	2.131066
7738	1975	029A	SOV	PL	QQ	MOLNIYA 3-2//29 NOV 1988	62.0	.2420	1.336683	11.026730
7741	1975	029D	SOV	RB	NA	MOLNIYA 3-2//SDC 7738	62.3	.7551	4.176679	1.996637
7780	1975	036A	SOV	PL	NI	MOLNIYA 1-29	63.2	.7410	4.163366	2.006173
7790	1975	038A	CAN	PL	NA	ANIK A3	6.0	.0002	6.621058	1.000451
7794	1975	038D	USA	RB	NA	ANIK A3//SDC 7790	24.6	.6448	2.932804	3.394490
7800	1975	036D	SOV	RB	NA	MOLNIYA 1-29//SDC 7780	63.7	.7289	4.221763	1.964718

Sid	Year	Designation	Owner	Mission	Status	Comment	Inclination	Eccentricity	Semi major axis	Mean motion
1 1 1 1 1	1 1 1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		1 1 1	-			1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		
7902	1975	042B	USA	82	NA NA	INTELSAT IVF-1//SDC 7815	26.1	.7216	3.911072	2.203979
7903	1975	049A	SOV	PL	DD	MOLNIYA 1-30//12 AUG 1987	62.8	.7267	3.731215	2.362655
8015	1975	063A	SOV	PL	NA	MOLNIYA 2-13	63.4	.7400	4.163128	2,006343
8018	1975	063D	Sov	828	NA	MOLNIYA 2-13//SDC 8015	63.8	.7293	4.221963	1.964575
8134	1975	077C	USA	RB.	KN	SYMPHONIE B//SDC 8132	13.7	TT27.	3.882609	2.228435
8187	1975	079A	Sov	PL	DD	MOLNIYA 1-31//19 NOV 1986	63.8	.6751	3.123115	3.087744
8195	1975	081A	SOV	PL	NI	MOLNIYA 2-14	63.1	.7452	4.163234	2.006262
8274	1975	079E	SOV	RB	DD	MOLNIYA 1-31//SDC 8187//08 JU	63.3	.7236	3.654255	2,439685
8331	1975	091B	USA	RB	KN	INTELSAT IVAF-1//SDC 8330	21.7	.7216	3.909765	2.205187
8366	1975	100A	USA	PL	AC	SMS C-GOES 1// SYNCTM 1120Z	9.6	. 0002	6.610121	1.002910
8418	1975	081D	SOV	RB	NA	MOLNIYA 2-14//SDC 8195	63.7	.7385	4.225118	1.962362
8425	1975	105A	SOV	PL	IN	MOLNIYA 3-3	63.3	.7404	4.163919	2.005770
8462	1975	1050	SOV	838	NA NA	MOLNIYA 3-3//SDC 8425	63.8	.7334	4.225243	1.962282
8476	1975	117A	USA	PL	AC	RCA Satcom 1	0.0	.0001	6.610865	1.002646
8482	1975	118A	USM	PL	AC	OPS 3165 MEWS 5	5.0	.0014	6.600486	1.005112
8492	1975	121A	SOV	PL	NA	MOLNIYA 2-15	62.8	.6703	3.089751	3.137869
8521	1975	125A	Sov	PL	QQ	MOLNIYA 3-4//12 AUG 1986	63.4	.6914	3.290762	2.854843
8529	1975	121D	SOV	RB	DD	MOLNIYA 2-15//SDC 8492//13 OC	63.8	.7319	3.760229	2.337269
8547	1975	123E	SOV	DB	KN	RADUGA 1//SDC 8513	46.5	.6410	2.857945	3.527971
8548	1975	049E	SOV	RB	Z.	MOLNIYA 1-30//SDC 7903	62.7	.7071	3.529642	2.570071
8585	1976	0048	CAN	PL	N.	CTS A	10.2	.0017	6.613727	1.002104
8600	1975	125F	Sov	828	DD	MOLNIYA 3-4//SDC 8521//30 JUL	63.8	.7178	3.572281	2.524123
8601	1976	0068	SOV	PL	N	MOLNIYA 1-32//TT	63.5	.7255	4.174432	1.998236
8620	1976	010A	ITS	PL	NA	INTELSAT IVAF-2	5.9	9000.	6.636881	.996858
8621	1976	010B	USA	RB	NA	INTELSAT IVAF-2//SDC 8620	22.0	.7202	3.913463	2.202021
8701	1976	006D	SOV	RB	NA	MOLNIYA 1-32//SDC 8601	63.6	.7280	4.076651	2.070525
8741	1976	021A	SOV	PL	NI	MOLNIYA 1-22	62.7	.7519	4.154772	2.012438
8751	1976	023F	USA	RB	NA	LES 8-9//SDC 8746-8747	20.1	.0140	6.700733	.982643
8762	1976	026A	SOV	PL	AC	MOLNYIA 1-34	64.0	.7263	4.162679	2.006672
8774	1976	029A	USA	PL	NA	RCA B (SATCOM II)	5.5	.0060	6.684373	.986265

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8820	1976	039A	USA	PL	NA	LAGEOS I	109.9	.0039	1.923765	6.386718
8822	1976	039C	USA	DB	NA	LAGEOS//SDC 8820	109.9	.0038	1.923371	6.388686
8833	1976	041A	Sov	PL	IN	MOLNIYA 3-5	62.2	.7399	3.910050	2.203367
8838	1976	042A	USA	PL	NI	COMSTAR 1-D1	5.7	.0002	6.631016	.998169
8840	1976	0428	USA	RB	NA	COMSTAR 1-D1//SDC 8838	21.6	.7159	3.891330	2.220879
8844	1976	041D	SOV	RB	NA	MOLNIYA 3-5//SDC 8833	61.9	.7536	4.133384	2.028069
8882	1976	053A	USA	PL	AC	MARISAT B (MARISAT 3)	7.4	.0002	6.610943	1.002729
8910	1976	053F	USA	RB	NA	MARISAT B (MARISAT 3) //SDC 88	25.4	.6807	3.265319	2.889287
8916	1976	059A	USM	PL	AC	OPS 2112 MEWS 6	0.8	. 0004	6.611030	1.002708
8918	1976	0590	USM	RB	NA	OPS 2112 MEWS 6//SDC 8916	5.	.0030	6.603496	1.004675
9007	1976	0658	USM	PL	AC	OPS 3986 SESP 74-2	97.4	.1940	1.283459	11.715864
6006	1976	066A	IND	PL	NA	PALAPA A	4.6	.0003	6.621068	1.000428
9017	1976	D990	USA	88	NA	PALAPA A//SDC 9009	24.6	6909	2.642385	3.969319
9047	1976	073A	USA	PL	AC	COMSTAR 1-D2	9.6	0000	6.610944	1.002703
9049	1976	074A	SOV	PL	QQ	MOLNIYA 1-39//TT//29 MAY 1987	63.4	.6826	3.212861	2.959299
9269	1976	074E	SOV	RB	NA	MOLNIYA 1-35//SDC 9049	62.6	.7276	4.032701	2.104541
9329	1976	073B	USA	RB	NA	COMSTAR 1-D2//SDC 9047	21.4	.7178	3.882406	2.228535
9330	1974	1016	USA	RB	NA	SYMPHONIE A//SDC 7578	13.0	.7303	3.933712	2.185157
9411	1976	021D	SOV	RB	NA	MOLNIYA 1-23//SDC 8741	63.0	.7383	4.214503	1.969832
9478	1976	101A	USA	PL	AC	MARISAT C (MARISAT 2)	0.6	.0001	6.611142	1.002685
9495	1976	105A	SOV	PĽ	NA	COSMOS 862//TT	67.1	.6965	4.162677	2.006607
9056	1976	1050	SOV	838	NA	COSMOS 862//SDC 9495	66.4	.6967	4.140154	2.023012
9574	1976	116A	SOV	PL	1N	MOLNIYA 2-16	62.1	.7434	4.163913	2.005853
9579	1976	116D	SOV	88	NA	MOLNIYA 2-16//SDC 9574	62.3	.7316	4.217925	1.967464
9635	1976	127A	SOV	PL	NA	MOLNIYA 3-6	59.5	.4918	2.099146	5.603660
9647	1976	127E	SOV	RB	QQ	MOLNIYA 3-6//SDC 9635//30 SEP	63.9	.5003	2.025750	5.910550
9829	1977	010A	SOV	PL	IN	MOLNIYA 2-17	63.7	.7167	4.163096	2.006418
9850	1977	O10E	SOV	RB	NA	MOLNIYA 2-17//SDC 9829	64.1	.6972	4.214698	1.969703
9852	1977	014A	JPN	PL	NA	ETS 2//KIKU 2	9.1	.0009	6.613187	1.002210
9880	1977	021A	SOV	PL	NI	MOLNIYA 1-36	63.7	.7241	4.163273	2.006281

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	1 1 1 1	1 1 1 1 1 1 1	1 1 1		1			1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	
9889	1976	105F	SOV	DB	NA	COSMOS 862//SDC 9495	67.0	9869.	4.165429	2.004620
9911	1977	027A	SOV	PL	NA	COSMOS 903	67.5	.6355	4.163897	2.005828
9921	1977	027D	SOV	RB	NA	COSMOS 903//SDC 9911	67.7	.6469	4.187609	1.988807
9927	1977	021D	SOV	RB	NA	MOLNIYA 1-30//SDC 9880	64.0	.7044	4.221640	1.964843
9931	1977	029A	ESA	PL	NA	ESRO-GEOS 1	27.6	.6604	4.226484	1.961763
9933	1977	029C	ESA	RB	NA	ESA-GEOS 1 //SDC 9931 (3RD ST)	26.0	.3153	1.514224	9.152373
9941	1977	032A	SOV	PL	IN	MOLNIYA 3-7	63.7	.7216	4.163304	2.006261
10000	1977	034A	USM	PL	NA	OPS 9437	9.3	.0019	6.774168	.966703
10001	1977	034B	USM	PL	NA	OPS 9438	6.8	9800.	6.832878	.954265
10002	1977	034C	USM	RB	NA	OPS 9437-9438//SDC 10000-1000	9.5	.0282	6.826526	.955599
10025	1977	0418	USA	RB	NA	INTELSAT IVAF-4//SDC 10024	21.2	.7206	3.890417	2.221650
10059	1977	047A	SOV	PL	NA	COSMOS 917	9.19	.5360	4.162015	2.007234
10001	1977	048A	USA	PL	AC	GOES B// SYNCTM 1415Z	7.9	.0032	6.600820	1.005040
10089	1977	047D	SOV	RB	KN	COSMOS 917//SDC 10059	67.4	.5570	4.181278	1.993376
10001	1977	053A	USM	PL	AC	NTS 2	64.6	.0052	4.164829	2.005211
10092	1977	054A	SOV	PL	NI	MOLNIYA 1-37	63.0	.7199	4.163270	2.006319
10143	1977	065A	JPN	PL	MA	GMS 1//HIMAWARI 1	8.1	.0015	6.656323	.992504
10150	1977	0688	SOV	PL	NA	COSMOS 931	67.2	.5751	4.162650	2.006765
10155	1977	052D	SOV	RB	NA	MOLNIYA 1-37//SDC 10092	63.5	.7066	4.076504	2.070701
10159	1977	071A	SOV	PL	IN	RADUGA 3	10.0	.0028	6.607962	1.003420
10167	1977	0680	SOV	RB	NA	COSMOS 931//SDC 10150	68.2	.5782	4.134114	2.027551
10315	1977.	082A	SOV	PL	NI	MOLNIYA 1-38	62.7	.7290	4.024304	2.111113
10369	1977	082E	SOV	RB	NA	MOLNIYA 1-38//SDC 10315	63.9	.7295	3.834542	2.269660
10422	1977	102E	USA	PL	QQ	ISEE A//26 SEP 1987	8.7	.9126	11.833726	.418747
10455	1977	105A	SOV	PL	IN	MOLNIYA 3-8	63.5	8069.	4.164077	2.005745
10485	1977	105E	SOV	RB	NA	MOLNIYA 3-8//SDC 10455	63.6	.6820	4.216482	1.968477
10489	1977	1088	ESA	PL	NA	METEOSAT 1	9.1	.0017	6.613294	1.002199
10516	1977	118A	JPN	PL	NA	CS//SAKURA	7.5	.0001	6.671310	.989153
10557	1978	002A	ITS	PL	NA	INTELSAT IVAF-3	4.1	.0003	6.626796	.999133
10605	1978	M600	SOV	PL	NA	MOLNIYA 3-9	63.5	.0612	1.076795	15.248947

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1 1 1		1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1					1 1 1 1 1 1 1 1 1 1 1 1			
10669	1978	016A	USM	PL	AC	FLTSATCOM 1//OPS 6391	8.2	.0003	6.611457	1.002604
10684	1978	020A	NSM	PL	AC	GPS 1//OPS 5111//NAVSTAR 1	63.7	.0116	4.164326	2.005591
10696	1978	024A	SOV	PL	NI	MOLNIYA 1-39	62.0	.7351	4.164045	2.005774
10722	1978	002B	USA	RB	NA	INTELSAT IVAF-3//SDC 10557	21.0	.7186	3.898747	2.214531
10723	1978	0120	USA	RB	NA	IUE//SDC 10637	29.5	.7194	3,743703	2.353406
10778	1978	035A	ITS	7d	NA	INTELSAT IVAF-6	4.2	8000	6.609938	1.002966
10779	1978	035B	USA	RB	NA	INTELSAT IVAF-6//SDC 10778	21.6	.7193	3.889729	2.222265
10794	1978	039C	JPN	RB	NA	BSE//SDC 10792	27.0	.4609	1.928681	6.365893
10801	1978	020B	USM	RB	NA	GPS 1//OPS 5111//SDC 10684	63.9	.4848	2.160974	5.364645
10802	1978	3600	Sov	RB	K	MOLNIYA 3-9//SDC 10609	63.6	.7367	3.871656	2.237114
10803	1978	024D	SOV	RB RB	NA	MOLNIYA 1-39//SDC 10696	62.2	.7369	4.206771	1.975288
10855	1978	044A	ESA	PL	AC	OTS 2	6.3	.0002	6.611729	1,002554
10893	1978	047A	USM	PL	Ac	GPS 2//OPS 5112//NAVSTAR 2	64.4	.0163	4.164458	2.005478
10894	1978	047B	USM	RB	NA	GPS 2//OPS 5112//SDC 10893	64.0	.5031	2.258748	5.020137
10925	1978	055A	SOV	PL	IN	MOLNIYA 1-40	63.3	.6918	4.163670	2.006045
10949	1978	OSSF	SOV	RB	NA	MOLNIYA 1-40//SDC 10925	63.5	.6898	4.220153	1.965909
10950	1977	108C	USA	RB	NA	METEOSAT//SDC 10489	27.1	.2215	1.317697	11.275614
10955	1978	062C	USA	RB	NA	GOES 3 (3rd Stage) //SDC 10953	23.7	.1873	1.260109	12.058846
10960	1977	053B	USA	RB	NA	NTS 2//SDC 10091	64.5	.5191	2.401229	4.580037
10970	1978	066A	Sov	PL	2	COSMOS 1024	67.8	.5486	4.163185	2.006375
10976	1978	068B	USA	RB	NA	COMSTAR D3//SDC 10975	22.0	.7160	3.893121	2.219335
10981	1978	071A	ESA	PL	NA	GEOS 2// SYNCTM 1535	0.6	.0001	6.650649	.993907
10983	1978	071C	USA	RB	NA	GEOS 2//SDC 10981	25.8	.6607	3.056156	3.190985
10984	1978	092A	SOV	PL	IN	MOLNIYA 1-41	62.0	. 7392	4.036654	2.101448
10987	1978	073A	SOV	PL	IN	RADUGA 4	9.8	.0011	6.607476	1.003522
10998	1978	0660	SOV	RB	NA	COSMOS 1024//SDC 10970	67.8	.5550	4.172478	1.999676
11007	1978	080A	SOV	PL	NA	MOLNIYA 1-42	63.8	.7213	4.163450	2.006160
11015	1978	083A	SOV	PL	NA	COSMOS 1030	9.79	.6031	4.161069	2.007874
11028	1978	087B	JPN	828	NA	EXOS B//SDC 11027	31.2	.6088	2.648467	3.955425
11054	1978	093A	USM	PL	AC	GPS 3//OPS 5113//NAVSTAR 3	63.9	.0062	4.164176	2.005687

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1 1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		1 1 1	1 1 1 1			1 1 1 1 1 1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
11057	1978	W560	SOV	PL	KN	MOLNIYA 3-10	63.1	.7034	4.164613	2.005360
11073	1978	072D	SOV	RB	NA	MOLNIYA 1-41//SDC 10984	62.4	.7449	4.074208	2.072444
11075	1978	0800	SOV	82	NA	MOLNIYA 1-42//SDC 11007	64.2	.7068	4.219564	1.966293
11076	1978	083D	SOV	RB	NA	COSMOS 1030//SDC 11015	6.59	.6226	4.184862	1.990802
11079	1978	3560	SOV	RB	NA	MOLNIYA 3-10//SDC 11057	63.2	.7109	4.226787	1.961273
11136	1978	1093	SOV	82	Z	COSMOS 1051-1058/SDC 11128-35	74.0	.0140	1.250081	12.189368
11141	1978	112A	USM	PL	AC	GPS 4//OPS 5114//NAVSTAR 4	63.7	6500.	4.164266	2.005628
11142	1978	1128	USM	82	NA	GPS 4//OPS 5114//SDC 11141	63.6	.5035	2.168458	5.336896
11144	1978	113A	USM	PL	AC	OPS 9441 DSCS 11	8.9	.0004	6.608731	1.003227
11145	1978	1138	USM	PL	AC	OPS 9442 DSCS 12	6.7	9000.	6.610930	1.002732
11153	1978	116A	CAN	PL	NA	ANIK B	3.5	9000.	6.630908	.998216
11158	1978	1187	SOV	PL	IN	GORIZONT 1	19.0	.3336	6.610774	1.002778
11240	1979	0048	SOV	PL	IN	MOLNIYA 3-11	63.9	.6845	4.164276	2.005596
11256	1979	A700	USA	PL	AC	SCATHA//OPS 7802//CHECK SYNCT	7.2	.1653	6.547909	1.017236
11273	1979	015A	SOV	PL	IN	EKRAN 3	9.1	.0038	6.613513	1.002139
11328	1979	031A	SOV	PL	QQ	MOLNIYA 1-43//09 DEC 1989	63.8	.3890	1.660161	7.966534
11353	1979	038A	USM	PL	AC	FLISATCOM 2//OPS 6392	6.9	8000.	6.610962	1.002737
11384	1979	0488	SOV	PL	IN	MOLNIYA 3-12	63.5	.7310	3.779156	2.319734
11436	1979	053C	USM	RB	NA	OPS 7484 MEWS 9//SDC 11397	1.5	.0057	6.648204	.994304
11440	1979	062A	SOV	PL	IN	GORIZONT 2	9.6	.0010	6.612037	1.002478
11474	1979	070A	SOV	PL	IN	MOLNIYA 1-44	63.9	.6755	4.164073	2.005753
11509	1979	A770	Sov	PL	NA	COSMOS 1124	68.1	.5850	4.159958	2.008682
11550	1979	0770	SOV	828	NA	COSMOS 1124//SDC 11509	68.1	.5907	4.186651	1.989503
11551	1979	031D	SOV	82	NA	MOLNIYA 1-43//SDC 11328	64.1	.7343	3.814105	2.287917
11553	1979	004D	SOV	88	NA	MOLNIYA 3-11//SDC 11240	64.2	.6738	4.222107	1.964541
11554	1979	0480	SOV	RB	NA	MOLNIYA 3-12//SDC 11384	63.9	.2619	1.382547	10.482293
11555	1979	058D	SOV	RB	NA	COSMOS 1109//SDC 11417	67.4	6099.	4.178737	1.995147
11556	1979	0700	SOV	RB	2	MOLNIYA 1-44//SDC 11474	64.2	.6723	4.222480	1.964282
11567	1974	017F	SOV	RB	NA	COSMOS 637//SDC 7229	11.2	.0047	6.578940	1.010059
11569	1976	107F	SOV	RB	NA	EKRAN 1//SDC 9503	10.3	.0012	6.559342	1.014575

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			1	1 1 1 1 1						
11570	1977	071F	SOV	RB	NA	RADUGA 3//SDC 10159	10.3	.0011	6.724728	. 977378
11571	1977	092F	SOV	RB	NA	EKRAN 2//SDC 10365	9.6	6000	6.566949	1.012815
11589	1979	091A	SOV	PL	NI	MOLNIYA 1-45	63.2	.7422	4.163954	2.005754
11602	1979	091D	SOV	RB	NA	MOLNIYA 1-45//SDC 11589	63.9	.7445	4.218978	1.966625
11621	- 6161	098A	USM	PL	AC	OPS 9443 DSCS 13 (Type 2)	6.3	.0001	6.610884	1.002734
11623	1979	098C	USM	RB	Ä	OPS 9443-9444//SDC 11622 AND	7.5	.0280	6.838063	.953190
11635	1979	1018	USA	PL	NA	SATCOM C	7.6	.4845	4.434398	1.825425
11661	1979	105D	SOV	DB	NA	GORIZONT 3 4TH STAGE FRACMENTS	46.6	.3902	1.538147	7.839730
11662	1980	002A	SOV	PL	IN	MOLNIYA 1-46	64.0	.7271	4.163677	2,005994
11670	1980	002F	SOV	RB	NA	MOLNIYA 1-46//SDC 11662	64.4	.7162	4.221082	1.965228
11676	1975	097F	SOV	RB	NA NA	COSMOS 775//SDC 8357	11.1	.0029	6.619022	1.000895
11684	1979	105E	SOV	RB	NA NA	GORIZONT 3//SDC 11648	8.4	.0016	6.681624	. 986858
11690	1980	011A	USM	PL	AC	GPS 5//OPS 5117//NAVSTAR 5	64.2	.0113	4.164337	2.005584
11705	1980	0118	USM	RB	NA NA	GPS 5//OPS 5117//SDC 11690	63.5	.5261	2.274022	4.969638
11708	1980	0168	SOV	PL	IN	RADUGA 6	8.5	.0018	6.611372	1.002642
11715	1980	0188	JPN	PL	NA	ECS 2//AYAME 2	6.3	.0002	6.609124	1.003142
11718	1980	018C	JPN	RB	Z.	ECS B//SDC 11715	24.4	.6008	2.598766	4.069713
11728	1980	016D	SOV	RB	NA	RADUGA 6//SDC 11708	8.7	.0016	6.729967	.976243
11758	1980	028A	SOV	PL	IN	COSMOS 1172	66.1	.6447	4.164949	2.005095
11762	1980	028E	SOV	RB	NA	COSMOS 1172//SDC 11758	66.7	.6610	4.180736	1.993731
11783	1980	032A	USM	PL	N C	GPS 6//OPS 5118//NAVSTAR 6	63.7	.0149	4.164278	2.005609
11791	1980	032B	USM	RB	NA	GPS 6//OPS 5118//SDC 11783	63.2	.4815	2.006176	5.997338
11844	1980	050A	SOV	PL	NA	COSMOS 1188	67.4	.6158	4.164143	2.005675
11847	1980	050B	SOV	RB	NA	COSMOS 1188//SDC 11844	67.4	.6149	4.182963	1.992157
11856	1980	053A	SOV	PL	IN	MOLNIYA 1-47	64.2	.7503	4.161057	2.007813
11861	1980	0530	SOV	82	NA	MOLNIYA 1-47//SDC 11856	64.6	.7487	4.221669	1.964740
11862	1980	049F	SOV	RB	Z.	GORIZONT 4//SDC 11841	8.1	.0028	6.715475	.979415
11871	1980	057A	SOV	PL	NA	COSMOS 1191	67.9	.5791	4.158619	2.009652
11888	1980	057D	SOV	RB	Z.	COSMOS 1191//SDC 11871	67.9	.5876	4.179726	1.994454
11896	1980	063A	Sov	PL	IN	MOLNIYA 3-13	63.6	.7226	4.163391	2.006174

Sid	Year	Designation	Owner	Mission	Status	Comment	Inclination	Eccentricity	Semi major axis	Mean motion
1 1 1 1	1 1 1 1		1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1					
11909	1980	063D	SOV	82	NA	MOLNIYA 3-13//SDC 11896	64.0	.7282	4.220178	1.965810
11926	1978	118C	SOV	RB	NA	GORIZONT 1//SDC 11158	18.9	.3315	6.553401	1.015970
11940	1973	0408	USM	R.B	NA	OPS 6157 MEMS 4//SDC 6691	7.6	.0041	6.642875	905566
11941	1978	073E	SOV	RB B	NA	RADUGA 4//SDC 10987	7.6	.0017	6.732621	.975658
11964	1980	074A	USA	PL	AC	GOES D	6.3	.0031	6.657232	. 992292
12032	1980	085A	SOV	PL	IN	COSMOS 1217	67.3	.6138	4.157970	2.010150
12035	1980	0850	SOV	RB	NA	COSMOS 1217//SDC 12032	67.5	.6333	4.179682	1.994493
12065	1980	M160	USA	PL	AC	SBS A//AKA AT3 3F3	2.9	.0013	6.611227	1.002678
12066	1980	092A	SOV	PL	NI	MOLNIYA 1-48	64.1	. 7333	4.148413	2.017025
12070	1980	092D	SOV	23	NA	MOLNIYA 1-48//SDC 12066	64.4	.7268	4.224519	1.962780
12078	1980	095A	SOV	PL	NA	COSMOS 1223	67.8	7765.	4.164952	2.005070
12086	1980	095E	SOV	82	NA	COSMOS 1223//SDC 12078	9.19	.6115	4.184700	1.990894
12120	1980	104A	SOV	PL	NI	EKRAN 6	7.9	.0003	6.608718	1.003252
12133	1981	002A	SOV	PL	NI	MOLNIYA 3-14	64.0	.6837	4.163580	2.006105
12134	1981	002B	SOV	RB	NA	MOLNIYA 3-14//SDC 12133	64.3	.6830	4.219042	1.966676
12156	1981	₹600	SOV	PL	NA	MOLNIYA 1-49	64.0	0569.	4.162963	2.006515
12159	1981	Q600	SOV	RB B	KN	MOLNIYA 1-49//SDC 12156	64.3	.7000	4.217012	1.968052
12295	1981	012A	JPN	PL	NA	KIKU 3//ETS 4	28.2	.6583	3.041901	3,213384
12303	1981	016A	SOV	PL	IN	COSMOS 1247	67.8	.6024	4.137721	2.024890
12309	1981	0188	USA	PL	AC	COMSTAR D4	4.1	.0001	6.611088	1.002697
12311	1981	016E	SOV	RB.	NA	COSMOS 1247//SDC 12303	9.19	9665.	4.108122	2.046815
12339	1981	025A	USM	PL	AC	OPS 7350 MEMS 11	. 1	9000.	6.611220	1.002665
12363	1981	0188	USA	RB	NA	COMSTAR D4//SDC 12309	20.1	.7168	3,896993	2,216042
12368	1981	030A	SOV	PL	NI	MOLNIYA 3-15	64.8	.7148	4.163061	2,006438
12371	1981	025C	NSM	RB	NA	OPS 7350 MEMS 11//SDC 12339	.1	.0001	6.564267	1.013442
12376	1981	031A	SOV	PL	NA	COSMOS 1261	6.79	.6154	4.162059	2.007147
12383	1981	030D	SOV	RB	NA	MOLNIYA 3-15//SDC 12368	65.0	.7070	4.221192	1.965147
12384	1981	031D	SOV	RB	NA	COSMOS 1261//SDC 12376	67.8	.6077	4.123651	2.035254
12445	1980	098B	USA	RB.	NA	INTELSAT VF-2//SDC 12089	23.8	.4670	1.964542	6.192570
12447	1980	081F	SOV	RB	NA	RADUGA 7//SDC 12003	8.1	\$000	6.624317	. 999703

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	1						1 1 1 1 1 1 1 1			
12471	1980	104E	Sov	R3	NA	EKRAN 6//SDC 12120//WAS 96117	7.8	.0037	6.564230	1.013452
12474	1981	050A	ITS	PL	AC	INTELSAT VF-1	2.0	.0003	6.611101	1.002685
12512	1961	054A	SOV	PL	IN	MOLNIYA 3-16	64.0	.6914	4.163742	2.005986
12519	1981	054E	SOV	RB.	NA	MOLNIYA 3-16//SDC 12512	64.3	.6873	4.224368	1.962961
12545	1981	057B	QNI	PL	NA	APPLE	6.3	9890.	6.400852	1.052511
12546	1961	057C	ESA	DB	NA	TECHNOLOGY CAPSULE FOR 12544	10.4	.7000	3.469175	2.638587
12547	1961	058A	SOV	PL	IN	COSMOS 1278	67.1	.6011	4.165105	2.004997
12556	1961	060A	Sov	PL	NI	MOLNIYA 1-50	64.6	.7343	4.163175	2.006326
12561	1961	058D	SOV	82	NA	COSMOS 1278//SDC 12547	67.5	.6170	4.187673	1.988804
12562	1961	057D	ESA	R3	NA	METEOSAT 2//SDC 12544	10.5	.5902	2,538886	4.214981
12563	1961	0090	SOV	82	NA	MOLNIYA 1-50//SDC 12556	6.19	.7301	4.218391	1.967074
12564	1961	061A	SOV	PL	IN	EKRAN 7	7.5	.0003	6.610073	1.002929
12627	1981	071A	SOV	PL	NA	COSMOS 1285	0.89	.6198	4.198788	1.980868
12677	1961	076A	JPN	PL	NA	GMS 2//HIMAWARI 2//MOVED 319	5.7	.0008	6.642737	.995550
12679	1961	070E	USA	82	NA	DE A//SDC 12624	0.68	.6219	2.873547	3.498237
12680	1961	071D	SOV	82	NA	COSMOS 1285//SDC 12627	0.89	.6152	4.182400	1.992522
12787	1981	012C	JPN	RB	NA	KIKU 3//SDC 12295	28.6	.7010	3.470236	2.637089
12810	1961	076C	JPN	RB	NA	GMS 2//SDC 12677	28.6	.3813	1.657350	7.991988
12815	1979	077F	SOV	DB	NA	COSMOS 1124//SDC 11509	64.3	.6458	4.128511	2.031681
12817	1979	077H	SOV	DB	NA	COSMOS 1124//SDC 11509	63.4	.6705	4.180975	1.993589
12818	1981	0888	SOV	PL	NA	COSMOS 1305	63.4	.4526	2.135577	5.460665
12827	1961	0888	SOV	82	NA	COSMOS 1305//SDC 12818	63.4	.4535	2.128496	5.487939
12833	1979	058E	SOV	DB	KN	COSMOS 1109//SDC 11417	67.4	.6586	4.152431	2.014135
12834	1979	058F	SOV	DB	NA	COSMOS 1109//SDC 11417	0.89	.5787	4.167778	2.003030
12850	1961	069F	Sov	82	NA	RADUGA 9//SDC 12618	7.6	.0024	6.726564	.976984
12851	1961	061F	SOV	82	NA	EKRAN 7//SDC 12564	7.4	.0002	6.578894	1.010077
12897	1961	102A	SOV	PL	IN	RADUGA 10	7.2	9000.	6.612061	1.002484
12906	1977	068E	SOV	DB	NA	COSMOS 931//SDC 10150	68.2	.5801	4.154967	2.012297
12907	1978	083E	SOV	DB	NA	COSMOS 1030//SDC 11015	64.1	.6361	4.138938	2.024023
12908	1978	016C	USM	DB	NA	FLISATCOM 1//SDC 10669	26.4	.4426	1.868724	6.674853

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12915	1981	1058	SOV	PL	NI	MOLNIYA 3-1	64.4	. 7172	4.147863	2.017457
12919	1978	0835	SOV	DB	NA	COSMOS 1030//SDC 11015	64.6	.6312	4.171604	2.000305
12920	1981	105E	SOV	RB	NA	MOLNIYA 3-17	64.5	.7078	4.222841	1.963987
12933	1981	1088	SOV	PL	NA	COSMOS 1317	67.0	.5952	4.168110	2.002788
12940	1981	108D	SOV	RB	NA	COSMOS 1317//SDC 12933	67.1	.6167	4.184377	1.991124
12959	1981	113A	SOV	PL	NI	MOLNIYA 1-51	64.0	.7022	4.163993	2.005799
12967	1981	114A	USA	PL	AC	RCA SATCOM III OR III-R//SYNC	τ.	. 0002	6.610797	1.002755
12984	1981	016F	Sov	DB	NA NA	COSMOS 1247//SDC 12303	67.8	.6013	4.135192	2.026747
12986	1981	113D	SOV	RB	NA NA	MOLNIYA 1-51//SDC 12959	64.3	.6934	4.090014	2.060459
12993	1981	071E	SOV	DB	NA	COSMOS 1285//SDC 12627	68.1	.6178	4.202086	1.978535
13001	1981	120E	SOV	PL	AC	RADIO 7	83.0	.0020	1.257734	12.080537
13007	1981	1198	USA	RB	NA	INTELSAT VF-3//SDC 12994	23.6	.4659	1.965853	6.186366
13011	1981	122B	ESA	DB	NA	TECH CAP + VIBRATION ISOL DEV	10.8	.7100	3.584479	2.512290
13012	1981	123A	SOV	PL	IN	MOLNIYA 1-52//TT	64.0	9604	4.163880	2.005823
13016	1981	1230	sov	R.B	N.A.	MOLNIYA 1-52//SDC 13012	64.2	. 6974	4.075956	2.071079
13025	1981	122C	ESA	RB	QQ	MARECS A//SDC 13010//21 NOV 1	10.3	.0886	1.119107	14.414567
13035	1982	0048	USA	PL	AC	RCA SATCOM IV	0.	.0002	6.611011	1.002717
13060	1982	3600	SOV	DB	NA	EKRAN 8//SDC 13056	46.7	.5051	2.067507	5.733829
13069	1982	0148	USA	PL	AC	WESTAR IV	0.	. 0004	6.610866	1.002753
13070	1982	015A	Sov	PL	IN	MOLNIYA 1-53	63.9	.7227	4.162944	2.006505
13075	1982	015D	Sov	RB	NA	MOLNIYA 1-53//SDC 13070	64.1	.7196	4.214036	1.970137
13080	1982	016A	Sov	PL	IN	COSMOS 1341	67.7	.6100	4.162928	2.006531
13089	1982	019B	USM	RB	NA	OPS 8701 IMEWS 13/SDC 13086	.7	.0012	6.573955	1.011194
13090	1982	016D	Sov	RB	NA	COSMOS 1341//SDC 13080	9.19	9609.	4.129090	2.031253
13091	1964	049F	Sov	DB	NA	COSMOS 41//SDC	71.1	.6719	4.150388	2.015556
13092	1982	020A	SOV	PL	NA	GORIZONT 5	6.9	.0035	6.688544	.985342
13098	1981	1148	USA	RB	NA	RCA/SATCOM 3R (PAM-D)	27.4	.7310	3.831213	2.272835
13112	1982	023D	SOV	DB	NA	MOLNIYA 3-18//SDC 13107	65.0	.7154	4.219604	1.966250
13124	1982	029A	SOV	PL	NA	COSMOS 1348	66.3	9909	4.168359	2.002602
13137	1982	031B	IND	RB	NA	INSAT 1A R/B (PAM-D)	28.5	.7220	3.683991	2.411778

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	1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1		1 1 1			1 1 1 1 1 1 1 1 1 1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1 1 1 1 1 1 1 1 1
13169	1982	0290	SOV	RB BB	NA	COSMOS 1348//SDC 13124	9.99	.6158	4.115358	2.041422
13177	1982	044A	SOV	PL	NI	COSMOS 1366	6.5	.0007	6.611606	1.002573
13205	1982	045A	SOV	PĽ	NI	COSMOS 1367	65.4	.6155	4.161064	2.007897
13215	1982	0450	SOV	82	NA	COSMOS 1367//SDC 13205	9.59	.6279	4.110326	2.045191
13237	1982	050A	SOV	PL	NA	MOLNIYA 1-54	64.8	.7059	4.163304	2.006276
13253	1982	OSOE	SOV	23	NA	MOLNIYA 1-54//SDC 13237	65.0	.7020	4.218973	1.966707
13269	1982	058A	USA	PL	AC	WESTAR V	0.	.0002	6.611020	1.002705
13294	1982	058B	USA	23	QQ	WESTAR V//SDC 13269//22 MAR 1	27.0	.0510	1.075291	15.300376
13295	1982	064A	SOV	PL	IN	COSMOS 1382	67.7	9609.	4.165718	2.004511
13298	1982	0640	SOV	RB BB	NA	COSMOS 1382//SDC 13295	9.19	.6195	4.127253	2.032591
13383	1982	074A	SOV	PL	NA	MOLNIYA 1-55	64.8	.7172	4.163213	2.006326
13390	1982	074D	SOV	RB	NA	MOLNIYA 1-55//SDC 13383	64.8	.7132	4.089145	2.061079
13431	1982	082A	CAN	PL	AC	ANIK DI	0.	.0001	6.610996	1.002712
13432	1982	083A	SOV	PC	IN	MOLNIYA 3-19	64.1	.7092	4.162830	2.006592
13446	1982	083E	SOV	23	NA	MOLNIYA 3-19//SDC 13432	64.4	.7153	4.222407	1.964264
13447	1982	082C	CAN	RB	NA	ANIK D1 (PAM-D) //SDC 13431	24.4	.7330	3.851909	2.254368
13554	1982	093A	SOV	PL	NI	EKRAN 9	5.7	.0023	6.609907	1.002960
13583	1982	0930	SOV	DB	NA	EKRAN 9 DEBRIS//SDC 13554	49.2	.6067	3.017529	3.251671
13585	1982	095A	SOV	PL	NA	COSMOS 1409	64.7	.6211	4.165949	2.004398
13591	1982	0950	SOV	8	NA	COSMOS 1409//SDC 13585	65.2	.6353	4.122668	2.036037
13595	1982	A760	ITS	PĽ	AC	INTELSAT VF-5	ĸ.	.0005	6.611012	1.002712
13603	1982	100A	SOV	PL	NA	COSMOS 1413	64.7	.0003	3.989694	2.138676
13606	1982	1000	SOV	PL	IN	COSMOS 1414	64.7	.0031	3.999207	2.131058
13607	1982	100E	SOV	PL	NA	COSMOS 1415	64.7	0000	3.990402	2.137989
13608	1982	100F	SOV	DB	NA	COSMOS 1413-1415	52.1	.5630	2.389078	4.615646
13609	1982	1000	SOV	DB	NA	COSMOS 1413-1415	52.1	.5685	2.436253	4.482226
13610	1982	100H	sov	RB	NA	COSMOS 1413-1415	64.7	.0007	3.987979	2.140066
13624	1982	103A	SOV	PL	NA	GORIZONT 6	6.1	.0013	6.606816	1.003680
13629	1982	103D	sov	DB	NA	GORIZONT 6 FRAGMENT//SDC 13624	46.8	.7238	3.694801	2.399914
13630	1982	103E	SOV	23	NA	GORIZONT 6//SDC 13624	6.1	.0023	6.605367	1.004003

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13631	1982	1058	USA	PL	AC	RCA SATCOM V	۲.	0000.	6.611096	1.002686
13637	1982	106B	USM	PL	AC	DSCS III//IRON 6451//SYNCTM 0	1.	.0004	6.610798	1.002768
13643	1982	106D	USM	RB	NA	IUS II//DSCS II-III	4.6	.0042	6.650169	.993882
13644	1982	103F	SOV	DB	NA	GORIZONT 6 FRAGMENT//SDC 13624	46.8	.6052	2.568867	4.125612
13651	1982	1108	USA	PL	AC	SBS C//AKA AT2 F2	1.	.0002	6.611063	1.002694
13652	1982	110C	CAN	PL	AC	ANIK C3	1.	.0004	6.610955	1.002739
13658	1982	1100	USA	RB	NA	SBS C//SBS F2//SDC 13651	22.9	. 7304	3.870926	2.238499
13666	1982	1105	CAN	RB	NA	ANIK C3//SDC 13652	22.7	.7306	3.867618	2.241377
13669	1982	113A	SOV	PL	IN	RADUGA 11	5.5	.0047	6.726116	.977085
13676	1982	113E	SOV	DB	NA	RADUGA 11 FRAGMENT//SDC 13669	46.6	.7004	3,393531	2.726618
13753	1976	023K	USA	DB	NA	LES 8-9//SDC 8746-8747	8.6	.0002	6.564141	1.013484
13782	1983	0068	JPN	PL	AC	CS 2A//SAKURA 2A MOVING FRO	2.6	0000	6.611113	1.002380
13875	1983	015A	Sov	PL	NA	MOLNIYA 3-20	64.0	7607.	4.160848	2.008033
13878	1983	016A	SOV	PL	IN	EKRAN 10	6.9	.0037	6.851947	.950304
13882	1983	015E	SOV	RB	NA	MOLNIYA 3-20//SDC 13875	64.0	.7056	4.218219	1.967220
13890	1983	A610	SOV	PL	IN	MOLNIYA 1-56	63.7	.6992	4.173638	1.998837
13897	1983	019D	SOV	RB	NA	MOLNIYA 1-56//SDC 13890	63.8	6969.	4.220774	1.965456
13899	1982	020F	SOV	RB	NA	GORIZONT 5//SDC 13092	6.9	.0048	6.683793	. 986389
13900	1979	015D	SOV	RB	NA	EKRAN 3//SDC 11273	0.6	.0008	6.564458	1.013398
13901	1983	020A	SOV	PL	NA	ASTRON	79.8	.7092	16.987947	.243381
13905	1967	0012	USA	DB	KN	INTELSAT IIF-1//SDC 2639	23.6	.4677	1.937389	6.323300
13907	1961	01AB	USA	DB	NA	INTELSAT IIF-2//SDC 2639	24.3	.6123	2.680806	3.884282
13908	1961	OIAC	USA	DB	NA	INTELSAT IIF-2//SDC 2639	28.4	.7060	3.556919	2.541279
13909	1961	01AD	USA	DB	NA	INTELSAT IIF-2//SDC 2639	25.6	.6765	3,323449	2.813828
13912	1961	01AG	USA	DB	NA	INTELSAT IIF-2//SDC 2639	26.8	.6422	2.914550	3.426348
13913	1969	0642	USA	DB	QQ	INTELSAT IIIF-5//SDC 4051//06	25.6	.0117	1.033708	16.235988
13914	1969	064#	USA	DB	KN	INTELSAT IIIF-5//SDC 4051//06				
13915	1969	64 AB	USA	DB	QQ	INTELSAT IIIF-5//SDC 4051//23	26.9	.0097	1.030101	16.319154
13939	1961	01AH	USA	DB	ž	INTELSAT IIF-2//SDC 2639	26.2	.1613	1.219524	12.665648
13940	1961	01AJ	USA	DB	NA	INTELSAT IIF-2//SDC 2639	24.3	.5442	2.275830	4.966209

Sid	Year	Designation	Owner	Mission	Status	Comment	Inclination	Eccentricity	Semi major axis	Mean motion
!	:		1	1 1 1 1						
13954	1982	113F	SOV	RB	NA	RADUGA 11//SDC 13669	5.7	.0028	6.732491	.975698
13958	1961	OLAL	USA	08	KN	INTELSAT 11F-2//SDC 2639	26.9	.6607	3.101599	3.121065
13960	1979	0583	SOV	08	NA	COSMOS 1109//SDC 11417	67.1	9899.	4.173199	1.999101
13961	1981	071F	SOV	DB	NA	COSMOS 1285//SDC 12627	64.3	9659.	4.198204	1.981329
13964	1983	025A	SOV	PL	NA	MOLNIYA 1-57//TT	64.1	.7130	4.157697	2.010325
13967	1983	025D	SOV	RB	NA	MOLNIYA 1-57//SDC 13964	64.3	.7002	4.091653	2.059193
13969	1983	0268	USA	PL	AC	TDRS A//STS-6	4.4	.0018	6.623135	. 999972
13970	1983	026D	USA	RB	NA	IUS 2//TDRS A//STS-6	2.7	.1850	5.499688	1.321523
13971	1983	026C	USA	RB	NA	IUS 1//TDRS A//STS-6	26.2	.7056	3.550078	2,548685
13974	1983	028A	SOV	PL	IN	RADUGA 12	5.3	.0004	6.610540	1.002837
13979	1983	028D	SOV	DB	NA	RADUGA 12 FRAGMENT//SDC 13974	46.9	.6628	3.014295	3.257041
13980	1983	028E	sov	08	NA	RADUGA 12 FRAGMENT//SDC 13974	47.4	.7259	3.844159	2.261466
13984	1983	030A	USA	PL	AC	RCA SATCOM VI OR I-R	1.	.0002	6.610953	1.002739
13990	1983	030C	USA	RB	QQ	RCA SATCOM VI//SDC 13984//27	24.1	.6372	2.827561	3.585827
14000	1977	5890	sov	08	KA	COSMOS 931//SDC 10150	65.7	.6268	4.166785	2.003732
14005	1979	062D	sov	RB	NA	GORIZONT 2//SDC 11440	8.8	.0005	6.727741	.976738
14034	1983	038A	SOV	PL	IN	COSMOS 1456	66.8	.6278	4.165452	2.004743
14050	1983	041A	USA	PL	AC	GOES F	2.1	.0003	6.610924	1.002745
14069	1983	041C	USA	82	NA	GOES F//SDC 14050	.3	.1594	7.419138	.843396
14070	1979	062E	SOV	DB	NA	GORIZONT 2 FRAGMENT//SDC 11440	46.5	.4626	1.905248	6.481507
14077	1983	047A	ITS	PL	AC	INTELSAT VF-6	. 2	.0004	6.611039	1.002699
14081	1983	0478	USA	RB	QQ	INTELSAT VF-6//SDC 14077//13	22.9	.0299	1.050765	15.842110
14095	1983	051A	ESA	PL	DD	EXOSAT//06 MAY 1986	74.9	. 9365	16.320628	.259120
14114	1982	044F	sov	RB	NA	COSMOS 1366//SDC 13177	9.1	.0127	6.610716	1.002779
14115	1982	093F	sov	RB	NA NA	EKRAN 9//SDC 13554	6.5	8000.	6.568056	1.012576
14117	1982	009F	sov	RB	NA	EKRAN 8//SDC 13056	6.9	. 0060	6.579804	1.009858
14130	1983	0590	ESA	RB	NA	ECS 1//OSCAR 10//SDC 14128	8.7	.6428	2.897747	3.456614
14134	1983	059C	INO	PL	AC	PALAPA B1//STS-7	т.	.0001	6.610899	1.002748
14158	1983	065A	USA	PL	AC	GALAXY I	т.	.0001	6.610764	1.002762
14166	1983	Q990	SOV	08	QQ	GORIZONT 7//SDC 14160//22 FEB	46.6	.7362	3.856019	2.251050

Sid	Year	Designation Owner	Owner	Mission	Status	Comment	Inclination	Eccentricity	Inclination Eccentricity Semi major axis	Mean motion
			1	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0				1 1 1 1 1 1 1 1 1 1
14168	1983	0650	USA	R3	KA	GALAXY I//SDC 14158	23.1	.6114	2.667348	3.913793
14182	1983	070A	SOV	PL	IN	COSMOS 1481	67.5	.6354	4.122797	2.035895
14190	1983	0728	USM	RB	NA	GPS 7//OPS 1204//SDC 14189	63.7	.5650	2.685356	3.872803
14193	1980	060F	SOV	828	NA	EKRAN 5//SDC 11890	8.1	.0016	6.553004	1.016065
14194	1981	027F	SOV	828	NA	RADUGA 8//SDC 12351	0.8	.0019	6.728101	.976659
14195	1981	102F	Sov	82	NA	RADUGA 10//SDC 12897	7.3	.0014	6.613034	1.002260
14206	1983	073D	SOV	RB	NA	MOLNIYA 1-58//SDC 14199	63.7	.0132	1.028085	16.344911
14258	1983	084A	Nos	PL	IN	COSMOS 1490	64.8	.0017	3.999237	2.131039
14259	1983	084B	Sov	PĽ	IN	COSMOS 1491	64.7	.0053	3.970319	2.154361
14260	1983	084C	SOV	77	NI	COSMOS 1492//RUN CASPER	64.8	.0001	4.003541	2.127612
14264	1983	084D	NOS	828	NA	COSMOS 1490-1492//SDC 14258-1	64.8	. 0005	4.001447	2.129274
14277	1983	084G	SOV	DB	NA	COSMOS 1490//SDC 14258	51.9	.5734	2.486381	4.347353
14287	1983	081C	JPN	RB	NA	CS 2B//SDC 14248	28.5	.5047	2.074515	5.706308
14313	1983	W060	SOV	PL	NI	MOLNIYA 3-21	64.5	.7218	4.159469	2.009002
14319	1983	Q060	SOV	82	MA	MOLNIYA 3-21//SDC 14313	64.4	T27.	4.215472	1.969098
14369	1983	0980	USA	RB	DD	GALAXY II//SDC 14365//31 DEC	23.7	.6238	2.734019	3.771445

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